

**DEPARTMENT OF TRANSPORTATION
FEDERAL AVIATION ADMINISTRATION**

1H13 Revision 9 California Helicopter Airways Inc. S62A May 06, 2015
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TYPE CERTIFICATE DATA SHEET NO. 1H13

This data sheet which is a part of type certificate No. 1H13 prescribes conditions and limitations under which the product for which the type certificate was issued meets the airworthiness requirements of the Civil Air Regulations.

Type Certificate Holder	California Helicopter Airways Inc. 279 Blackland Road Royse City, TX 75189
Type Certificate Holder Record	Sikorsky Aircraft transferred TC 1H13 to California Helicopter Airways Inc. on May 06, 2015

I - Model S-62A (Amphibious Transport Helicopter), approved June 30, 1960.

Engine	General Electric CT58-100-1, or CT58-100-2, or CT58-110-1 or CT58-110-2. See engine nameplate and/or General Electric Bulletin CT58 No. 45 and revisions thereto for appropriate derated fuel control part numbers.			
Fuel	Aviation Kerosene JP4 or JP5 (General Electric Co. Spec. No. F9134A).			
Engine limits	Sea level static			
Turbine		Shaft	Power	Power
		HP	Turbine R.P.M.	Gas Gen. R.P.M. Inlet (T5)°F
	Takeoff	730	20960 (100%Nf)	26300 (100%NG) 1165
	Maximum continuous	671	20960 (100%Nf)	25600 (97%NG) 1105
	Maximum transient			1250
	Starting			1200
	Takeoff and Maximum continuous horsepower ratings are normally obtained at power turbine speed of 20340 r.p.m. (97%Nf) and 19500 r.p.m. (93%Nf) respectively.			
Rotor Limits	Maximum 245 r.p.m. (Dual tachometer reading 107%) Minimum 170 r.p.m. (Dual tachometer reading 75%)			
Airspeed limits (IAS)	Vne (never exceed) 109 m.p.h. (95 knots) (See NOTE 6).			
Center of Gravity (C.G.) range	(+249.0) to (+262.0)			
Empty weight Center of Gravity C.G. range	None			
Datum	252 in. forward of main rotor centroid.			
Leveling means	Plumb line from top of main cabin door frame.			
Maximum weight	7500 lb. (See NOTE 6 for higher weight).			
Maximum Crew	1 (Pilot)			

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Maximum Passengers	19, limited by emergency exit requirements.
Maximum Baggage	None (See NOTE 3 for cargo).
Fuel capacity	187 gal. (+252)
Oil capacity	2.5 gal. (+174) See NOTE 1 for data on system fuel and oil.
Rotor Blade and Control Movements	For rigging information, refer to Maintenance Manual.
Serial Nos. eligible	62001 through 62003, 62005 through 62007, 62009 through 62016, 62020, 62027, 62028, 62031, 62033, 62058 through 62060, 62062, 62063, 62078, 62079, 62080, 62081, 62105, 62106, 62107, 62108, 62121, 62122, 62123, 62124.
Certification Basis	CAR 7, August 1, 1956 (Category B) including Amendment 7-1, 7.203 and 7.612(2)(4) of Amendment 7-4, 6.246 of Amendment 6.2 to CAR 6, December 20, 1956, and Exemption No. 100 dated June 30, 1960. Type Certificate No. 1H13 issued June 30, 1960. Date of Application for Type Certificate January 31, 1958.
Production Basis	Production Certificate No. 105.
Equipment	The basic required equipment as prescribed in the applicable airworthiness regulations (see certification basis) must be installed in the aircraft for certification. In addition the following items of equipment are required: (a) FAA Approved Rotorcraft Flight Manual.

NOTE 1. Current weight and balance report including list of equipment included in certificated empty weight, and loading instructions when necessary, must be provided for each helicopter at the time of original certification. The certificated empty weight and corresponding C.G. locations must include undrainable oil of 0.8 lb. (+174) and unusable fuel of 21 lb. (+252).

NOTE 2. The following placard must be displayed in front of and in clear view of the pilot:
"THIS HELICOPTER MUST BE OPERATED IN COMPLIANCE WITH THE OPERATING LIMITATIONS SPECIFIED IN THE FAA APPROVED ROTORCRAFT FLIGHT MANUAL."

NOTE 3. The cabin floor area between stations 180 and 348 is structurally satisfactory for a uniformly distributed loading of 175 p.s.f. when used for cargo purposes, with the following limitations. The areas between stations 276 and 312 and stations 312 to 348 are each limited to a maximum of 1200 lb. The maximum total floor load permitted is 2400 lb.

NOTE 4. Information essential to the proper maintenance of the helicopter including retirement time of critical components is contained in the Sikorsky S62A Maintenance Manual provided with each helicopter. Sikorsky publication No. SA4045-22 revision dated March 26, 1982, Maintenance Manual Model S62A helicopter, is applicable to the Sikorsky Model S62A helicopter. The values of retirement or service life cannot be increased without FAA engineering approval.

NOTE 5. The cargo sling and hoist are special purpose equipment and are to be operated in accordance with the limitations described in FAR 133. Additional operating limitations are also contained in revised Rotorcraft Flight Manual Supplements No. 3 (September 27, 1972), 6 (September 27, 1972), and 7 (September 27, 1972).

NOTE 6. The S62A is eligible for a gross weight of 7900 lb. when modified in accordance with Sikorsky Kit Drawing S6207-45100. Serial Nos. 62021 and up have been modified by the manufacturer in production. Supplement No. 10 dated July 3, 1963, to the FAA Approved Rotorcraft Flight Manual is required. The Vne (never exceed) speed is 101 m.p.h. (88 knots) IAS.

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