

**DEPARTMENT OF TRANSPORTATION
FEDERAL AVIATION ADMINISTRATION**

A00001AC Revision 5 MT, LC SJ30-2 May 20, 2011

TYPE CERTIFICATE DATA SHEET NO. A00001AC

This data sheet which is part of Type Certificate No. A00001AC prescribes conditions and limitations under which the product for which the type certificate was issued meets the airworthiness requirements of Title 14 of the Code of Federal Regulations.

Type Certificate Holder	MT, LC 1770 Skyplace Blvd, Ste. 100 San Antonio, TX 78216
Type Certificate Holder Record	Type Certificate initial issuance to Sino Swearingen Aircraft Corporation on October 27, 2005 Sino Swearingen Aircraft Corporation transferred to Emivest Aerospace Corporation (same address) on October 16, 2008 Emivest Aerospace Corporation transferred to MT, LC (same address) on April 21, 2011

1 - Model SJ30-2, (Commuter Category), Approved October 27, 2005

Engines Two Williams-Rolls, Inc. International FJ44-2A Turbofans

Fuel Commercial kerosene JET A, JET A1, per ASTM-D1655, or JP-8 per MIL-T-83133.

Fuels not containing icing inhibitors must have MIL-I-27686 or MIL-I-85470 fuel system icing inhibitor added in amounts not less than 0.10% nor more than 0.15% by volume.

Dupont Stadis 450 anti-static additive or equivalent is permitted to bring fuel up to 300 conductive units, but not to exceed 1 part per million.

SOHIO Biobor JF biocide additive is approved at a concentration not to exceed 20 parts per million (270 ppm total additive) of elemental boron.

Engine Limits Static thrust standard day, sea level

Takeoff (5 minutes) static thrust, sea level	2,300 lbs
Maximum continuous static thrust, sea level	2,300 lbs

Max permissible engine rotor operating speeds (Takeoff and Maximum Continuous):

Low pressure rotor, N1 (30 seconds)	106.4%
Low pressure rotor, N1	105.2%
High pressure rotor, N2	98.8%

All other engine limits as noted in engine TCDS E3GL.

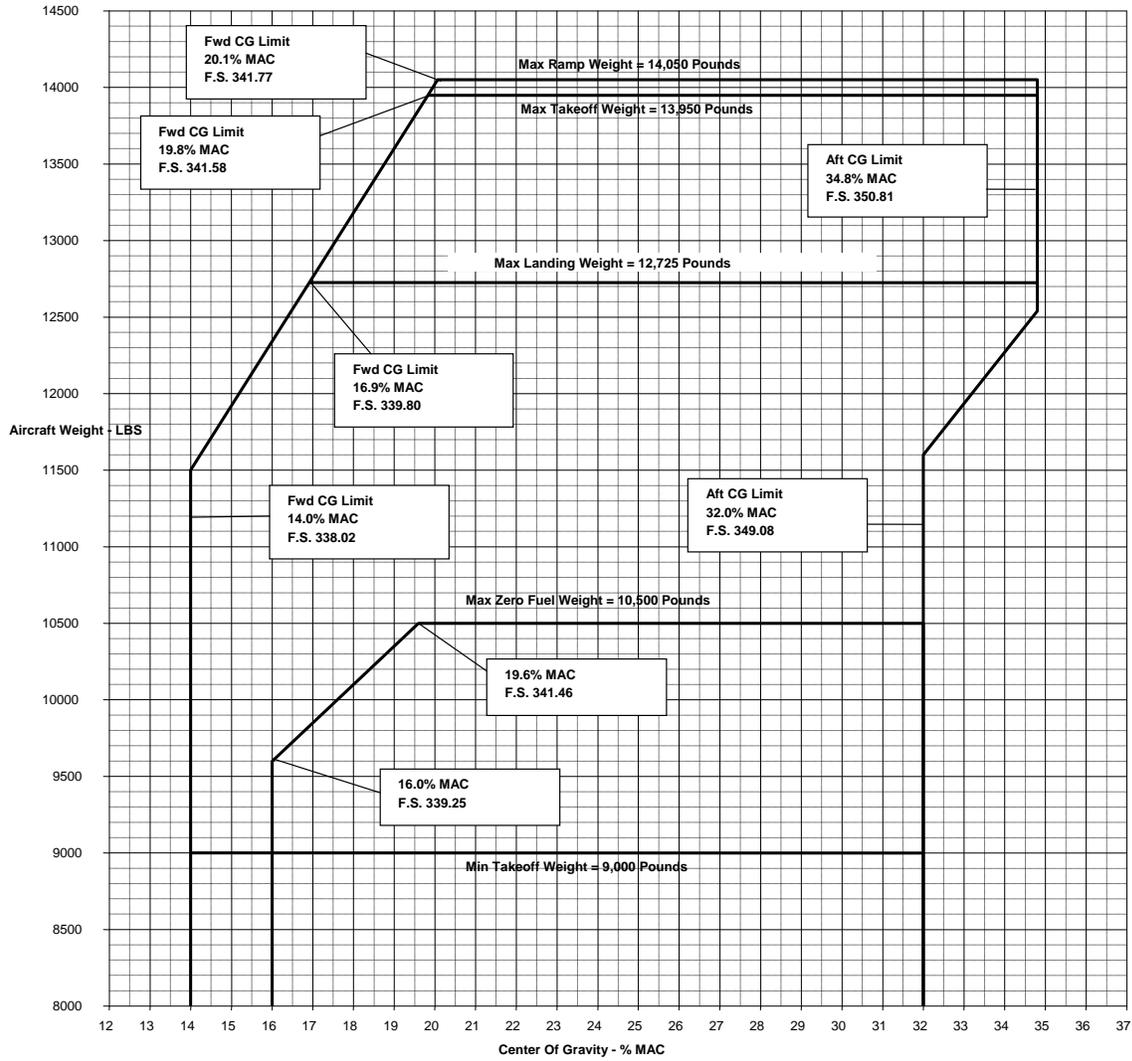
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Rev 4: Changed Max tire speed to 165 knots and added 4 aircraft serial numbers to Note 7 re RVSM compliance.

Airspeed limitations

V_{MO} (Maximum Operating Speed)	
Sea level to 29,500 feet	320 KCAS
above 29,500 feet M_{MO} (Maximum Operating Mach No.)	0.83
V_A (maneuvering speed at sea level)	255 KIAS (255 KCAS)
<i>See AFM for variations with altitude.</i>	
V_{FE} (Flaps extended)	
10 degrees	200 KIAS (199 KCAS)
20 degrees	200 KIAS (199 KCAS)
31 degrees (landing)	170 KIAS (169 KCAS)
V_{MCA} (Minimum control speed) Air	
Takeoff	79 KCAS
Landing	76 KCAS
V_{MCG} (Minimum control speed) Ground	85 KCAS
V_{LO} (landing gear operating)	
V_{LO}	225 KIAS
$V_{LO(EMER)}$	160 KIAS
V_{LE}	225 KIAS
Maximum autopilot operating speed	Any normal operating speed
Maximum tire ground speed	165 knots

Center of Gravity Range



Forward Limits: 14% MAC from 8,000lbs to 11,500lbs. Linear variation from 14% MAC (11,500lbs) to 20% MAC at 14,050 MAC

Aft Limits: 32% MAC from 8,000lbs to 11,700lbs. Linear variation from 32% (11,700lbs) to 34.75% MAC (12,500lbs). 34.75% MAC from 12,500lbs to 14,050lbs

All CG data is with landing gear extended and flaps/slats retracted.

Empty Wt. C.G. Range None

Datum	FS 0.00 is located 86.01 inches forward of the nose of the aircraft.								
Mean Aerodynamic Chord	61.48 inches. The leading edge of the mean aerodynamic chord is 329.41 inches aft of the datum.								
Leveling Means	Leveling Means: Locate leveling tool at FS 295.50 ± 2.00” for longitudinal and lateral leveling.								
Maximum weights	<table> <tr> <td>Takeoff</td> <td>13,950 lbs.</td> </tr> <tr> <td>Landing</td> <td>12,725 lbs.</td> </tr> <tr> <td>Zero Fuel</td> <td>10,500 lbs.</td> </tr> <tr> <td>Ramp</td> <td>14,050 lbs.</td> </tr> </table>	Takeoff	13,950 lbs.	Landing	12,725 lbs.	Zero Fuel	10,500 lbs.	Ramp	14,050 lbs.
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Ramp	14,050 lbs.								
Minimum Crew for all Flights (see Note 5 for cockpit equipment/arrangement restrictions):	<p>One pilot (in the left pilot seat) plus additional equipment as specified in the Kinds of Operations Equipment List (KOEL) contained in the Limitations Section of the FAA Approved Airplane Flight Manual</p> <p>OR</p> <p>One pilot and one copilot</p>								
Number of Seats	<p>Maximum seven (See AFM for loading instructions)</p> <p>Pilot and copilot (or passenger) at FS 188.0</p> <p>Side facing seat at FS 216.000</p> <p>Two aft facing seats at FS 278.275</p> <p>Two forward facing seats at FS 326.500</p>								
Maximum Baggage	Aft baggage 500 lbs (See AFM for loading instructions)								
Fuel Capacity (Usable)	713 U.S. GAL (4850 lbs at 6.8 lb./gal.) – Arm 352.62 (See Note 1 for unusable)								
Oil Capacity	3.27 Quarts usable per engine – Arm 452.50								

Maximum Operating Altitude	49,000 feet (flaps and gear retracted)	
	18,000 feet (flaps or gear extended)	
Control Surface Movements		
Rudder	Right	27.5° ± 0.5° hingewise
	Left	27.5° ± 0.5°
Rudder Trim	Right	38.5° ± 2.5° hingewise
	Left	38.5° ± 2.5°
Ventral Rudder	Right	30.0° ± 1° hingewise
	Left	30.0° ± 1°
Elevators	Up	22.0° ± 0.5° streamwise
	Down	17.2° ± 0.5°
Horizontal Tail Incidence	L. E. Up	1.7° +0.4° / -0.2° (L.E. Up 1.5° to 2.1°)
	L. E. Down	14.3° ± 0.5°
Ailerons	Neutral	1.0° +0.25°/-0.0 Down
	Up	15.5° ± 0.5° hingewise
	Down	11.3° ± 0.5°
Wing Flap/Slat	Cruise	UP, Multiple tolerances
	Takeoff	10°, 20°, Multiple tolerances
	Landing	LDG, Multiple tolerances
Speedbrake	Extended	30° ± 2° hingewise

All Control Surface movements are in accordance with the Instructions for Continued Airworthiness.

Manufacturers Serial Numbers 005 and up

Certification Basis – SJ30-2:

14 CFR part 23, effective February 1, 1965, as amended by Amendments 23-1 through 23-55 (3/1/02) including §23.562 for all seat places.

14 CFR part 36 effective December 1, 1969, as amended by Amendments 36-1 through 36-26 (8/4/05).

14 CFR part 34 as amended by Amendments 34-1 through 34-3 (11/30/04).

Title 49 U.S.C. Section 44715.

Special Conditions as follows:

23-ACE-87; additional requirements for:

HIRF, performance, takeoff, takeoff speeds, accelerate-stop distance, takeoff path, takeoff distance and takeoff run, takeoff flight path, climb, climb one engine inoperative, landing, bailed landing, stall speed, trim, static longitudinal stability, demonstration of static longitudinal stability, static directional and lateral stability, stall demonstration, stall characteristics, stall warning, vibration and buffeting, high speed characteristics, flight flutter testing, out-of-trim characteristics, pressure vessel integrity, fasteners, landing gear, ventilation, air conditioning, pressurization, airspeed indicating system, static pressure system, oxygen equipment and supply, maximum operating limit speed, minimum flight crew, airplane flight manual, operating limitations, operating procedures, and performance information.

23-105-SC; additional requirements for:

Side-facing lavatory seat.

Exemptions as follows:

No. 6742 for certification in the commuter category.

No. 6791 for landing gear loads from 14 CFR 23.25, 23.29, 23.235, 23.471, 23.473, 23.477, 23.479, 23.481, 23.483, 23.485, 23.493, 23.499, 23.723, 23.725, 23.726, 23.727, 23.959, 23.1583(c)(1) and (2), Appendix c23.1, Appendix D23.1.

Equivalent Level of Safety Findings as follows:

No. ACE-98-3 on emergency exit dimensions (14 CFR 23.783(f)(1))

No. ACE-01-02 for digital only N₂ and fuel flow display (14 CFR 23.1305)

No. ACE-05-17 ELOS for 1-g stall criteria March 4, 2004, 14 CFR 23.69(a)(4), 23.69(b)(5), Additional Method of Compliance for 23.143, 23.145(a),(b)(1) thru (b)(5), 23.147(a), 23.149(b), 23.157(b)(4), 23.233(a), 23.729(a)(1), (a)(2), 23.735(a)(2), 23.1001(b)(1), (b)(3), 23.1323 (b)(1), (b)(2), 23.1325(e), 23.1545(b)(3), (b)(4), Special Conditions 23-ACE-87, dated 31 October 1997: Special Condition 4(b)(1), 1st Additional Method of Compliance for 4, 2nd Additional Method of Compliance for 4, Special Condition 6(a), Special Condition 10(c), 10(d), 10(d)(3), 10(d)(4), Special Condition 11(a)(2), Special Condition 12(b), Special Condition 13(a), (a)(1), (a)(2), (a)(3), (a)(4), (b), (b)(1), (b)(2), Additional Method of Compliance for 13(a), Additional Method of Compliance for 13, Special Condition 14(b), (c)(1), (c)(2)(i), (c)(3), (d), Special Condition 16(a)(2), (b)(1), (b)(2), (b)(3), (c)(4), (b)(2)(ii), (c), (d), (d)(5), Special Condition 17(a), (b)(1), Special Condition 18(a)(2), Special Condition 20(c), 1st Additional Method of Compliance for 20, 2nd Additional Method of Compliance for 20, Special Condition 39(b)(2).

No. ACE-05-16 ELOS requested for airspeed indicator markings. (14 CFR 23.1545 (b)(4))

No. ACE-05-15 ELOS requested for storage battery design and installation. (14 CFR 23.1353(h))

No. ACE-05-25 ELOS requested for Ditching Provisions and Emergency Egress (14 CFR 23.807(e)(1))

No ACE -05-26 ELOS requested for SJ30-2 Main Cabin Door Opening (14 CFR 23.783(f)(1))

Compliance with ditching provisions has been met for issuance of a Type Certificate.

Compliance with ice protection for flight into known or forecast icing has been demonstrated for issuance of a Type Certificate.

Type Certificate: A00001AC, issued October 27, 2005

Date of application: July 2, 2003

Model SJ30-2 is defined by SSAC drawing 30-00001, Revision A dated 5/1/06, or later FAA approved revision.

Production Basis

None. Before original airworthiness certification of each aircraft, an FAA representative must perform a detailed inspection for workmanship, materials, conformity with the approved technical data, and a check of the flight characteristics.

In the event of an application for a standard airworthiness certificate or, if an applicant intends to produce a new aircraft under 14 CFR § 21.183(d), and the applicant is manufacturing, building, or

assembling to another person's type certificate, the applicant must provide written evidence of permission from the type certificate holder. Conduct of such activity without written evidence of permission may be a violation of 49 U.S.C. § 44704(a)(3)".

Serial Numbers

The following Serial Numbers were produced under 14 CFR Subpart F- Production Under Type Certificate Only: 006 thru 008 and 010.

Aircraft Serial Numbers 005 thru 007 were manufactured by Sino Swearingen Aircraft Corporation under Production Under Type Certificate Only.

Aircraft Serial Numbers 008 and 010 were manufactured by Emivest Aerospace Corporation using an Approved Production Inspection System.

Equipment

The basic required equipment as prescribed in the applicable airworthiness regulations (see Certification Basis and MMEL) must be installed in the aircraft for certification. (Refer to Limitations Section of FAA Approved Airplane Flight Manual for Kinds of Operation Equipment List)

"Additional or special equipment necessary for type certification (IAW Certification Basis)"
Autopilot, Yaw Damper and Rudder Bias are required to be operational for single pilot operation.

NOTES

Note 1 A current weight and balance report, including a list of the equipment included in the certification empty weight and loading instructions when necessary, must be provided for each aircraft at the time of original airworthiness certification.

The certificated empty weight and corresponding center of gravity location must include:

Unusable fuel (undrainable)	20 lbs at Arm 382.5
Unusable fuel (drainable)	53 lbs at Arm 307.0
Full Oil	16 lbs at Arm 452.5
Full Hydraulics	9 lbs at Arm 393.2

Note 2 Airplanes must be operated according to 30-030, "Airplane Flight Manual (AFM)", SJ30-2 FM-01, dated July 14, 2006, or later approved revision.
Placards as defined by drawings 30-81100, 30-90032, 30-95500 and 30-96002 must be installed.

Note 3 Airworthiness Limitations: 30-080, "SJ30-2 AIRCRAFT MAINTENANCE MANUAL", Chapter Four (4), "Airworthiness Limitations" for mandatory component retirement life information, required inspections and inspection intervals.

Note 4 All replacement seats or alterations of existing seats (crew and passenger) must meet the requirements of 14 CFR §23.562.

Note 5 Approval for operation with a minimum crew of one pilot is based upon the cockpit equipment installation and arrangement evaluated during FAA certification testing. No significant changes may be made to the installed cockpit equipment or arrangement (EFIS, autopilot, avionics, etc.), except as permitted by the approved MMEL, without prior concurrence from the responsible Aircraft Certification Office.

Note 6 Model SJ30-2 airplanes are approved for high altitude operations (altitudes above 41,000 feet), by Special Conditions. Any modifications to the pressure vessel must be approved in accordance with the requirements as shown in the certification basis.

Note 7 Airplane Serial Numbers identified below meet the airworthiness requirements for operation in Reduced Vertical Separation Minimum (RVSM) airspace.

S/N SJ30-2 005, 006, 007, 008, and 010 meet the airworthiness requirements for operation in Reduced Vertical Separation Minimum (RVSM) airspace. Until group approval is obtained, each aircraft must receive individual RVSM airworthiness and operational approval directly from the FAA.

Note 8 The SJ30-2 shall not be eligible for operations under 14 CFR Part 121

Note 9 The 30-91220-1 or 30-91220-3 toilet installation must be installed in order for the Side Facing Seat at FS 216.000 to be occupied for Taxi, Take-off, and Landing.

Note 10 `Model SJ30-2airplanes are approved with a 9” aisle width below 25” from the floor based on substantiation testing (emergency egress evacuation tests in accordance with 14 CFR §23.803(a)).

--END--