

**DEPARTMENT OF TRANSPORTATION
FEDERAL AVIATION ADMINISTRATION**

H1GL Scheutzow Model B May 4, 1976

TYPE CERTIFICATE DATA SHEET NO. H1GL

This data sheet which is a part of type certificate No. H1GL prescribes conditions and limitations under which the product for which the type certificate was issued meets the airworthiness requirements of the Federal Aviation Regulations.

Type Certificate Holder Scheutzow Helicopter Corporation
Columbia Station, Ohio

I - Model B Helicopter

Engine Lycoming IVO-360-A1A

Fuel 100/130 minimum grade aviation gasoline

Engine limits Take-off (5 min.), 2900 rpm, 28.3 in. Hg. (180 hp)
Max. continuous, 2900 rpm, 27.7 in. Hg. (175 hp)
See FAA approved Rotorcraft Flight Manual for variation of
maximum manifold pressure with ambient conditions.

Rotor limits and engine operating speeds Power Off (Rotor Tach.) Power On (Engine Tach.)
Minimum 400 rpm Minimum 2750 rpm
Maximum 485 rpm Maximum 2900 rpm

Airspeed limits (IAS)

VARIATION OF V _{ne} WITH ALTITUDE AND TEMPERATURE										
PRESSURE ALTITUDE IN FEET										
OAT °F	0	1000	2000	3000	4000	5000	6000	7000	8000	9000
-20	93	93	93	93	93	93	89	84	79	76
0	93	93	93	93	93	89	83	78	74	71
20	93	93	93	93	89	83	78	74	70	66
40	93	93	92	89	83	78	74	70	67	-
60	93	90	86	83	79	74	70	67	-	-
80	90	86	83	79	74	70	67	-	-	-
100	86	83	79	74	70	66	-	-	-	-

INDICATED AIRSPEED - MPH (with zero instrument error)

Altitude limits Take-off and landing, 6300 ft. density altitude maximum; enroute, 8700 ft. density altitude maximum.

C.G. range
 longitudinal (+96.0) to (+100.5)
 lateral (+1.0) right to (-1.4) left

Maximum weight 1735 lbs.; refer to Approved Rotorcraft Flight Manual for variation of weight with altitude

No. of seats 2 (+65.3) longitudinal, (+13.5) and (-9.5) lateral

Maximum baggage 100 lbs. (+87) longitudinal; lateral is optional

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Fuel capacity	21 gal. total, (+99.1) longitudinal, (-17.4) lateral Moment arms shown are for full tank. See FAA Approved Flight Manual for variation of moment arms with fuel quantity. (See Note 1 for data on unusable fuel)
Oil capacity	6 qts. total, (+115) longitudinal, (See Note 1 for data on unusable oil)
Rotor blade and control movements	Refer to Maintenance Manual for rigging information
Stabilizer setting	Refer the Maintenance Manual
Serial Nos. eligible	104 and up
Datum, longitudinal: lateral:	100 in. forward of center line of main rotor mast Airframe centerline or center line of mast
Leveling means	Rotor mast (vertical)
Certification basis	Part 27 of Federal Aviation Regulations, effective February 1, 1965 and Amendments 27-1 through 27-9 Application for Type Certificate dated March 13, 1975
Production basis	None - Type Certificate only
Equipment:	The basic required equipment as prescribed in the applicable airworthiness regulations (see certification basis) must be installed in the helicopter for certification. In addition, an FAA Approved Rotorcraft Flight Manual is required.

NOTE 1. Current weight and balance report including list of equipment in certificated empty weight, and loading instructions when necessary, must be provided for each helicopter at the time of original certification. The certificated empty weight and corresponding center of gravity locations must include unusable fuel of 0.3 gal. at (104.5) longitudinal, (-20.5) lateral and unusable oil of 1.5 qts. at (+115) longitudinal, (0) lateral.

NOTE 2. The following placard must be displayed in front of and in clear view of the pilot:

"This helicopter must be operated in compliance with the operating limitations specified in the Approved Rotorcraft Flight Manual and Maintenance Manual Airworthiness Limitations."

See FAA Approved Rotorcraft Flight Manual for additional operational limitations and placards.

NOTE 3. Prior to the issuance of any Standard Certificate of Airworthiness, an acceptable Rotorcraft Maintenance Manual must be furnished to the FAA by the manufacturer.

NOTE 4: Information essential for proper maintenance of the helicopter is contained in the Maintenance Manual. The retirement times of critical parts are listed in Section 1 titled, Critical Parts, Airworthiness Limitations, and Scheutzow Helicopter Corporation Report No. SHC 592, Revision F, Retirement Life of Critical Components Scheutzow Model B Helicopter, dated April 5, 1976.

Values of retirement times of critical parts cannot be increased without FAA approval.

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