

FEDERAL AVIATION ADMINISTRATION

2A12 Revision 1 CLARK 12 January 3, 1968
--

TYPE CERTIFICATE DATA SHEET NO. 2A12

This data sheet which is a part of type certificate No. 2A12 prescribes conditions and limitations under which the product for which the type certificate was issued meets the airworthiness requirements of the Civil Air Regulations.

Type Certificate Holder Clark Aircraft, Inc.
 1704 East Crockett Street
 Marshall, Texas

I - Model 12, 1 POLB (Restricted Category), Approved June 28, 1960

Engine	Lycoming R-680-B4 Series (See NOTE 3 for optional engine)		
Fuel	80 minimum grade aviation gasoline		
Engine limits	For all operations, 2100 r.p.m., 225 hp.		
Propeller and propeller limits	McCauley 41D5926/SS135-6 Diameter: Not over 102 in., Not under 100 in. No further reduction permitted. Static r.p.m. at maximum permissible throttle setting: Not over 1925, Not under 1800 No additional tolerance permitted.		
Airspeed limits	Never exceed 117 m.p.h. (102 knots) True Ind. Maneuvering 90 m.p.h. (78 knots) True Ind.		
C.G. range	(+8.6) to (+10.8)		
Empty weight C.G. range	None		
Datum	Leading edge of lower wing Leveling means Top of fuselage just aft of firewall		
Maximum weight	3340 lb.		
No. seats	1 (+63)		
Maximum cargo	1000 lb. (+9)		
Fuel capacity	46 gal. (-3) (45 gal. usable) See NOTE 1 for data on fuel system.		
Oil capacity	4.4 gal. (-33)		
Control surface movements	Elevator	Up 24 degrees	Down 18 degrees
	Elevator tab	Up 22 degrees	Down 22 degrees
	Rudder	Right 30 degrees	Left 30 degrees
	Aileron	Up 24 degrees	Down 23 degrees
Serial Nos. eligible	12-1 and up		
Certification basis	CAR 8 dated October 11, 1950 and CAM 8 dated September 1956 including all supplements 1 thru 18 dated March 10, 1958. Type Certificate No. 2A12 issued June 28, 1960. Date of Application for Type Certificate March 13, 1959.		

Page No.	1	2
Rev No.	1	1

Reformattd 2/94.

Production basis None. Prior to original certification of each aircraft an FAA representative must perform a detailed inspection for workmanship, material, and conformity with the approved technical data, and a check of flight characteristics.

Equipment The basic required equipment as prescribed in the applicable airworthiness regulations (see certification basis) must be installed in the aircraft for certification.

NOTE 1. Current weight and balance report including list of equipment included in the certificated empty weight and loading instructions when necessary must be in each aircraft at the time of certification and at all time thereafter except in cases of operators having an approved weight control system.

The certificated empty weight and corresponding center of gravity location must include unusable fuel 6 lbs. (-3).

NOTE 2. The following placards must be displayed:

(a) In clear view of the pilot:

"THIS AIRPLANE MUST BE OPERATED IN ACCORDANCE WITH THE FOLLOWING LIMITATIONS:

Maximum Speed 117 MPH
Maximum gross weight 3340 lbs.
No acrobatic maneuvers including spins authorized.
C.G. range (+8.6) to (+10.0) at 3340 lbs."

(b) On the hopper compartment:

"MAXIMUM HOPPER CAPACITY 1000 LBS."

(c) Adjacent to the fuel shut-off valve:

"USABLE FUEL 45 GALS. 80 OCTANE."

(d) On tachometer, when Continental W-670 engines installed:

"AVOID CONTINUOUS OPERATION BETWEEN 1500 AND 1650 RPM."

NOTE 3. Optional Engine Installation

Engine Continental W-670-6A or W-670-6N
Fuel 80 minimum grade aviation gasoline
Engine limits For all operations, 2075 r.p.m. (220 hp.)
Propeller and McCauley hub D-1093 and blades
propeller limits SS-138-6 or hub 41D5926 and blades SS-135-6
This propeller must be indexed in the zero degree position (blades in line with the crankthrow).
Diameter: Not over 102 in., not under 100 in.
No further reduction permitted.
Static r.p.m. at maximum throttle setting:
Not over 1900, Not under 1850