

Ignition system	Bendix-Scintilla ignition unit, low voltage capacitor discharge type, P/N 10/80375-10 with two Bendix-Scintilla P/N 10-102792-1 spark ignitors.
Certification basis	Type Certificate No. 1E6 issued June 16, 1960 Date of type certificate application August 29, 1957.
Production basis	None. The manufacturer does not hold a production certificate for the production of engines under this certificate and, therefore, each engine so produced is subject to a detailed inspection for workmanship and conformity with the approved data by a Federal Aviation Agency representative. In addition, the engine must have a satisfactory run-in including at least five hours at rated power and speed. Upon satisfactory completion of the above, the representative will tag the engine with Tag Form 186.

NOTE 1. Maximum permissible engine operating r.p.m., for the engine rotors are as follows:
Compressor Turbine, r.p.m. 24,440
Power Turbine, r.p.m. 20,370

NOTE 2. Maximum permissible temperatures are as follows:
Turbine exhaust gas temperatures at the tail pipe:
Takeoff (5 minutes) 1045°F (563°C)
Maximum continuous 1000°F (538°C)
Maximum for acceleration 1400°F (760°C) for 5 seconds
1200°F (649°C) for remainder of the transient time
Starting (5 seconds) 1400°F (760°C)
Oil inlet temperature 190°F (88°C)

NOTE 3. Fuel and oil pressure limits:
Fuel, at engine inlet 14.7 p.s.i. a min. TO
50.0 p.s.i.g. max.
Oil, at engine inlet 0 to 15 p.s.i.g.
Oil pressure
at idle 10 p.s.i.g. min.
operating range 70-90 p.s.i.g.

NOTE 4. The engine ratings are the guaranteed minimums and are based on static sea level conditions as follows:
Compressor inlet air 59°F and 29.92 in. Hg.
Inlet nozzle TE 516
Exhaust diffuser extension EXP-1287 with an exit area of 203 sq. in.
Exhaust inner cone assy. EXP-1799
No aircraft accessory loads
No compressor air bleed
No anti-icing airflow

NOTE 5. The following accessory drive provisions are incorporated:

Drive	Rotation (C-Clockwise	Gear Ratio	Max. Torque (in. - lb.)		Maximum Overhang Moment (in.-lb.)
			Cont.	Static	
<u>Gas Producer</u>	<u>CC-Counter-clockwise</u>				
Tachometer	C	.167	7	50	--
Starter-Generator	C	.283	150	1290	500
Spare	C	.147	68	330	25
<u>Power Turbine</u>					
Tachometer	CC	.695	7	50	--

NOTE 6. This engine meets the Federal Aviation Administration requirements for adequate turbine disc integrity and rotor blade containment and does not require external armoring. This engine has demonstrated satisfactory operation in icing conditions as defined in 4B.1(b) 7 and 8.

NOTE 7. The engine is not eligible for installation in certificated aircraft until approved installation, operation, maintenance, and overhaul manuals are available.

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