

**DEPARTMENT OF TRANSPORTATION
FEDERAL AVIATION ADMINISTRATION**

R00002RD
REVISION 2
AGUSTA
AB139
AW139

November 28, 2006

TYPE CERTIFICATE DATA SHEET NO. R00002RD

This data sheet which is a part of Type Certificate No.R00002RD prescribes conditions and limitations under which the product for which the Type Certificate was issued meets the airworthiness requirements of the Federal Aviation Regulations.

Type Certificate Holder

AGUSTA S.p.A.
Via Giovanni Agusta, 520
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I. Model AB139 (S/N 31001 through S/N 31054) and Model AW139 (S/N 31055 onwards). See NOTE 9.

(Transport Helicopter: Category A, Approved April 10, 2006; Category B, Approved December 20, 2004.)

Engine. Two (2) Pratt and Whitney Canada Inc. PW PT6C-67C, Type Certificate No. E00068EN
Free turbine turboshaft engines with FADEC.

Fuel. For all temperatures:
Jet A-1

Oil. For all temperatures:

MIL-PRF-83282 Hydraulic Oil
Alternative for low temperatures MIL-PRF-5606

MIL-PRF-23699 Transmission Oil

For engine oils see Engine Maintenance Manual

For detailed information see Section I of the approved Rotorcraft Flight Manual .

Installed Engine Limits.

	Max torque % (Lbft)	Max ITT °C	Max gen Speed rpm	Max Output Shaft speed rpm
OEI 2 ½ min	160% (400)	835	40500	21000 (21420)*
OEI Continuous	140% (350)	775	39100	21000 (21420)*
Take Off 5 min	110% (275)	775	39100	21000 (21420)*
Maximum continuous	100% (250)	735	38200	21000 (21420)*

(*) For Category-A take off / landings below 90 KIAS and for external load operations.

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Transmission limits.

MCP Max Continuous OEI
 2.5 min OEI
 MCP Max Continuous AEO
 TOP Take-Off AEO

Power @ 100% NR
 hp
 1400
 1600
 1000 (x 2)
 1100 (x 2)

Rotor Limits.

Power OFF: 90% < Nr < 116%
 Power ON: 95% < Nr < 106%

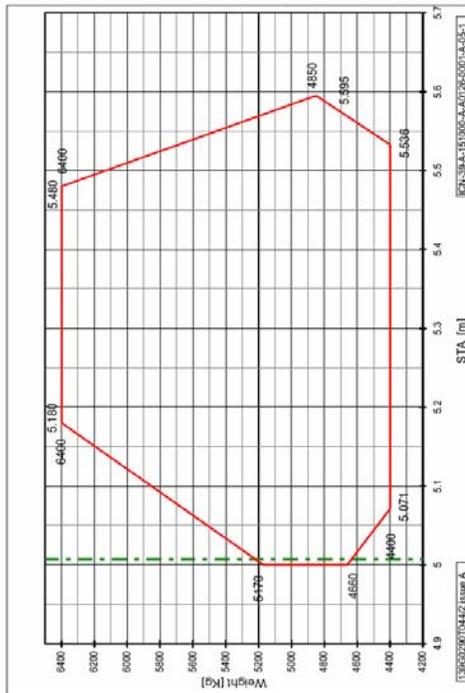
Air Speed Limits.

Vne Power On = 167 KIAS. See approved Rotorcraft Flight Manual for variations with density altitude.

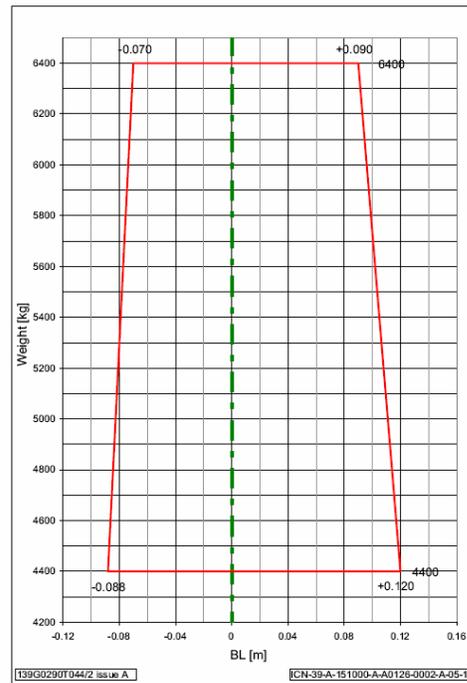
VLE/VLO (gear extended/gear operating) = 150 KIAS/150 KIAS

OEI/Power Off = 147 KIAS. See approved Rotorcraft Flight Manual for variations with density altitude.

Center of Gravity (C. G.) Range:



Weight And Longitudinal Cg Envelope

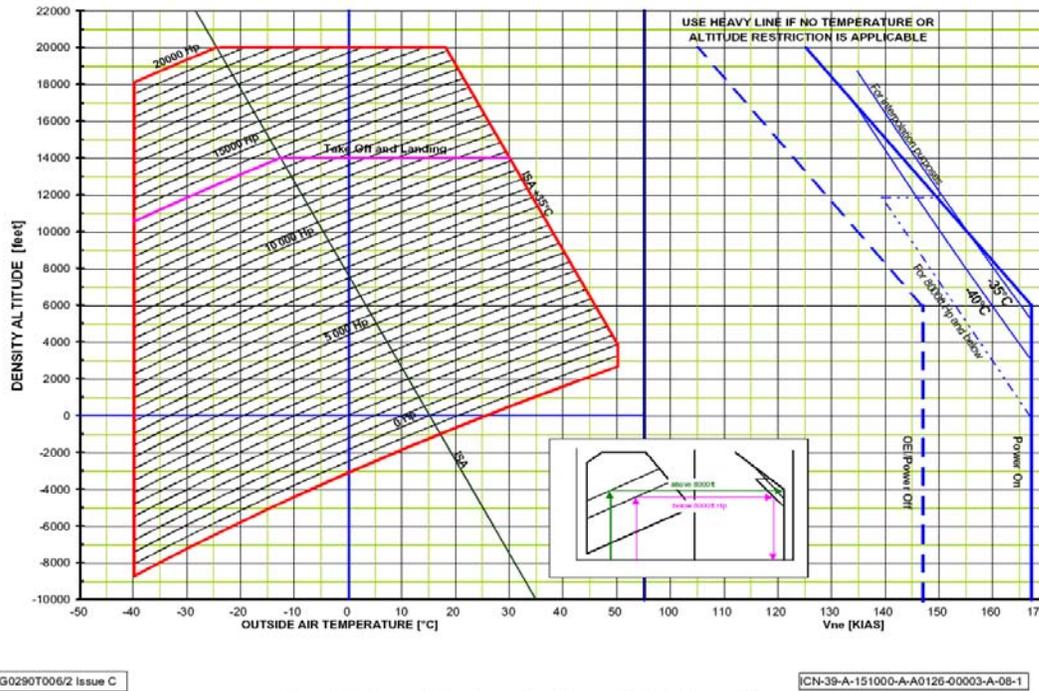


Weight And Lateral Cg Envelope

Empty weight Center of Gravity Range.

None

Ambient Temperature Limitations.



Datum.

Longitudinal station 0 (datum) is 3160 mm forward of the front jack point. Lateral station 0 (datum) is ± 905 mm inboard of each main jack point and coincides with the rotorcraft longitudinal plane of symmetry.

Levelling Means.

Plumb line from ceiling reference point to index plate on floor of passenger cabin located at STA=5055 mm and BL=750 mm (LHS).

Maximum weights.

- (Taxi and ramp) 6400 kg
- (Take-off) 6450 kg
- (Landing) 6400 kg

Minimum Crew.

Two (2) for VFR/IFR (See NOTE 5)

Number of Seats.

17 (2 crew – 15 passengers maximum)

Maximum Baggage.

200 kg (440 lb)

Baggage compartment max pressure load 300 kg/m² (61 lb/sq.ft)

Baggage compartment max tied up load 200 kg (440 lb)

Fuel Capacity.

Total: 1588 liters
Unusable for tank: 10 liters

Engine Oil Capacity (single).

8.45 U.S. Quarts (8.0 liters)

Maximum Altitude for take off and landing.

14,000 ft

Maximum Operating Altitude.

20,000 ft

Rotor Blades and Control Movements.

For rigging information, refer to the AB139/AW139 Maintenance Manual.

Import Requirements.

To be considered eligible for operation in the United States, each aircraft manufactured under this Type Certificate must be accompanied by a Certificate of Airworthiness for Export or certifying statement endorsed by the exporting foreign civil airworthiness authority which states the following (in the English language):

“The rotorcraft covered by this certificate has been examined, tested and found to conform to the type design approved under FAA Type Certificate No. R00002RD and to be in condition for safe operation.”

The only aircraft eligible for import into the United States are those aircraft with the 4-displays configuration defined in Agusta report No. 139G0000P005/02, “AB139 Type Design Definition – 4 Displays Configuration,” Rev. B, dated Nov. 15, 2004, or later approved revision. Agusta report 139G0000P005/02 is section 2 of the basic Agusta report 139G0000P005, “AB139- Type Design Definition,” Rev. D, dated November 15, 2004. (See NOTE 5).

Certification Basis:

- (1) 14 CFR Part 21.29 and Part 29, Amendment 29-1 through 29-45.
- (2) Appendix B to Part 29 of Amendment 29-40
- (3) 14 CFR 36, Appendix H, Amendment 36-1 through Amendment 36-25
- (4) Special Condition is made in accordance with 14 CFR 21.16 as follows:
Special Condition No. 29-0010-SC, High Intensity Radiated Fields (HIRF), dated Feb. 19, 2004.
- (5) Equivalent Level of Safety Findings issued against:
 - (a) 14 CFR 29.1305, as documented in AB139 FAA Memo dated Dec. 20, 2004.
 - (b) 14 CFR 29.1321, as documented in AB139 FAA Memo dated Dec. 20, 2004.

Equipment.

The basic required equipment as prescribed in the applicable airworthiness regulations (See Certification Basis) must be installed in the helicopter for certification.

See Agusta Report 139G040W002 - Equipment List

The installation of the following is mandatory for Category-A operations:

- Service Bulletin P&WC S.B. 41020
- Honeywell Primus EPIC software P/N MM7030191-004 or later approved software.

Refer to approved Rotorcraft Flight Manual for other approved mandatory and optional equipment.

Service information.

Agusta Service bulletins, structural repair manuals, vendor manuals, aircraft flight manuals, and overhaul and maintenance manuals, which contain a statement that the document is (ENAC/EASA) approved, are accepted by the FAA and are considered faa approved. These approvals pertain to the approved type design ONLY.

Flight Manual.

ENAC/EASA approved Rotorcraft Flight Manual (4-Display Configuration), 139G0290X002, dated November 25, 2004, or later approved revision (See NOTE 5).

Maintenance Manual.Maintenance Planning Information 39-A-AMPI-00-P Maintenance Publication
39-A-AMP-00-P.

NOTES

NOTE 1 Current weight and balance report including loading instructions and list of equipment included in the certificated empty weight, must be provided for each helicopter at the time of original airworthiness certification.

NOTE 2 All placards indicated in the approved Rotorcraft Flight Manual must be installed in the appropriate location.

NOTE 3 Information essential to the proper maintenance of the helicopter is contained in the Manufacturer's Maintenance Manual provided with each helicopter. Life limited components and associated retirement times are presented in Chapter 4 and must be replaced in accordance therewith.

NOTE 4 The model AB139/AW139 rotorcraft employs electronic engine controls, commonly named Full authority Digital Engine Controls (FADEC), that are recognized to be more susceptible to Electromagnetic Interference (EMI) than rotorcraft that have non-electronic controls. (EMI may be the result of radiated or conducted interference.)
For this reason modifications that add or change systems that have the potential for EMI, must either be qualified to a standard acceptable to the FAA or tested at the time of installation for interference with the FADEC. This type of testing must employ the particular FADEC's diagnostic techniques and external diagnostic techniques. The test procedure must be FAA approved.

NOTE 5 The FAA Rotorcraft Flight Manual (RFM) is identical to the ENAC/EASA approved RFM with exceptions for Minimum Flight Crew limitations and identification of Noise Characteristics. Unique FAA approved pages are marked as "FAA Approved" and must be included in the FAA manual to reflect the differences noted below:

1. Section 1, LIMITATIONS, MINIMUM FLIGHT CREW:
Requires Two pilots for VFR and Two pilots for IFR
2. Section 4, Performance Data, NOISE CHARACTERISTICS:

Model: AB139 Engine Pratt and Whitney PT6C-67C Gross Weight 6400 kg					
Configuration	Level Flyover EPNL (EPNdB)	Take Off EPNL (EPNdB)		Approach EPNL (EPNdB)	
		100 % NR	102 % NR	100 % NR	102 % NR
Clean aircraft No external kits Installed	89.5	90.4	90.9	92.7	93.2

NOTE 6 The Agusta Model AB139/AW139 incorporates an integrated avionics system using software-based line replaceable units (LRU) which share a digital signal transmission bus. The software configuration of the AB139/AW139, as delivered from production, is critical to the proper operation of the avionics and cockpit instrumentation system. Modification to the LRU software supplied with the AB139/AW139, replacement of an LRU with a different LRU, addition of new LRU, or alteration of an LRU interface could adversely affect the airworthiness of the certified software. No changes to the integrated avionics system should be made without coordination with the FAA Aircraft Certification Office (ACO) having jurisdiction over the modifier.

NOTE 7 The hydraulic fluids must conform to MIL-PRF-83282 or MIL-PRF-5606 which is an alternate for low Temperature operation - see LIMITATIONS Section of the approved Rotorcraft Flight Manual. The Landing Gear Shock Absorber must be filled only with MIL-PRF-5606.

- NOTE 8 Any changes to the type design of this helicopter by means of a amended type certificate (TC), supplemental type certificate (STC), or amended STC, requiring instructions for continued airworthiness (ICA's) must be submitted thru the project aircraft certification office (ACO) for review and acceptance by the Fort Worth -Aircraft Evaluation Group (FTW-AEG) Flight Standards District Office (FSDO) prior to the aircraft delivery, or upon issuance of the first standard airworthiness certificate for the affected aircraft, whichever occurs later as prescribed by Title 14 CFR 21.50. Type design changes by means of a field approval that require ICA's must have those ICA's reviewed by the field approving FSDO.
- NOTE 9 The AB139 and AW139 are two names of the same product. They identify two production batches manufactured in conformity with the same Type Design. Model AB139 includes S/N 31001 through S/N 31054. Model AW139 includes S/N 31055 onwards. Where not specifically declared, the content of this TCDS is applicable to both the AB139 and AW139.

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