

Airspeed Limits (IAS)	Maneuvering Maximum structural cruising Never exceed Landing gear operating Landing gear extended Flaps extended - takeoff Flaps extended - landing Minimum control	160 KIAS 212 KIAS 241 KIAS 182 KIAS 182 KIAS 182 KIAS 152 KIAS 78 KIAS
C.G. Range (Landing Gear Extended)	(+170.31) to (+179.08) at 8400 lb. (+165.6) at 6100 lb. or less Straight line variation between points given Landing gear retracted moment change: +1113 in./lb.	
Empty Wt. C.G. Range	None	
Leveling Means	Two screws located on W.L. 93.80 @ sta. 248.25 and sta. 272.65	
Maximum Weight	<u>S/N 404-0001 through 404-0200</u> Landing 8100 lb., takeoff 8400 lb. <u>S/N 404-0201 and up</u> Landing 8100 lb., ramp 8450 lb., takeoff 8400 lb.	
No. of Seats	<u>One through eleven</u> (2 at +137.0, 2 at +171.0, 2 at +199.0, 2 at +227.0, 1 at +255.0 and 2 at +296.0) See manufacturer's equipment list for other seating arrangements	
Maximum Baggage	250 lb. (+32.0), 350 lb. (+71.0), 400 lb. (+211.0), 400 lb. (+301.0) and 100 lb. (+317.0)	
Fuel Capacity	<u>S/N 404-0001 through 404-0200</u> 348.0 gallons (2 wing tanks, 174.0 gallons each, 170.0 gallons usable at +181.2) <u>S/N 404-0201 and up</u> 348.0 gallons (2 wing tanks, 174.0 gallons each, 172.0 gallons usable at +181.2) See NOTE 1 for data on unusable fuel	
Oil Capacity	26 quarts (13 quarts in each engine at +129.0; usable 7.0 quarts per engine) See NOTE 1 for data on undrainable oil.	
Maximum Operating Altitude	30,000 feet	
Control Surface Movements	Wing flaps	Down 35°, ±1° (Inboard) Down 23°, +0°, -1° (Outboard)
	Main surfaces	
	Aileron Up 25°, +1°, -0°	Down 15°, +1°, -0°
	Elevator Up 24°, +1°, -0°	Down 15°, +1°, -0°
	Rudder Right 32°, +1°, -0°	Left 32°, +1°, -0°
	(Read degrees normal to rudder hinge line)	
	Tab (main surface in neutral)	
	Aileron Up 15°, +1°, -0°	Down 15°, +1°, -0°
	Elevator Up 4°, +0.5°, -0.0°	Down 14°, +0.0°, -0.5°
	Rudder Right 9°, +0.0°, -0.5°	Left 9°, +1°, -0°
	(Read degrees normal to rudder hinge line)	
Serial Nos. Eligible	404-0001 through 404-0859	

II - Model 406, Caravan II, (Normal Category), Approved June 27, 1986

Engines	Two Pratt and Whitney Aircraft of Canada, Ltd., PT6A-112 turboprops																										
Fuel	Aviation turbine fuel ASTM D-1655, Jet A, Jet A-1, or Jet B; MIL-T-5624, JP-4, JP-5; MIL-T-83133, JP-8. Anti-icing additive per MIL-I-27686D, MIL-I-27686E, or Phillips PFA55MB must be blended into the aircraft fuel in concentrations not less than 0.060% or more than 0.15% by volume. For emergency use of aviation gasoline and fueling procedures, refer to approved Airplane Flight Manual.																										
Engine Limits	<u>Operating Limits</u>																										
	Shaft Horsepower	Ng Gas Generator Speed (% r.p.m.)	Indicated Torque (ft.-lbs.)	Prop Shaft Speed (r.p.m.)	Maximum Permissible Interturbine Temp. (°C)																						
Takeoff static and max. continuous	500	101.6	1382	1900	725																						
Starting (2 seconds)	-	-	-	-	1090																						
Maximum reverse	480	101.6	1382	1815	725																						
Propeller and Propeller Limits	Two McCauley three-bladed, full-feathering, reversible Hub: 3GFR34C701 Blade: 93KB-0 Diameter: Not over 93 inches, not under 90-5/8 inches; no further reduction permitted Pitch at 30-inch station: Low pitch 18.5°, feathered 85.5°, reverse -13.5°																										
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C.G. Range (Landing Gear Extended)	(+166.99 in.) to (+180.28 in.) at 6,500 lb. or less (11% to 32% MAC) (+172.42 in.) to (+180.28 in.) at 9,360 lb. (19.6% to 32% MAC) Variation is linear between points Landing gear retracting moment (+1346 in.-lb.)																										
Empty Wt. C.G. Range	None																										
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Maximum Weight	<table border="0" style="width: 100%;"> <tr> <td>Takeoff</td> <td style="text-align: right;">9,360 lb.</td> </tr> <tr> <td>Landing</td> <td style="text-align: right;">9,360 lb.</td> </tr> <tr> <td>Zero fuel (with zero wing locker payload)</td> <td style="text-align: right;">8,500 lb.</td> </tr> <tr> <td>Ramp</td> <td style="text-align: right;">9,435 lb.</td> </tr> </table>					Takeoff	9,360 lb.	Landing	9,360 lb.	Zero fuel (with zero wing locker payload)	8,500 lb.	Ramp	9,435 lb.														
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Number of Seats	<u>One through fourteen</u> (2 at +137.0, 2 at +166.0, 2 at +192.0, 2 at +218.0, 2 at +244.0, 2 at +270.0, and 2 at +296.0) See manufacturer's equipment list for other seating arrangements																										

Maximum Baggage	250 lb. (+32.0), 350 lb. (+71.0), 400 lb (+211.0), 400 lb. (+301.0), and 100 lb. (+317.0)		
Fuel Capacity	3227 lb. (481.5 gal.) total in two wing tanks, 1613.5 lb. (240.75 gal.) each 3183 lb. (475 gal.) usable total, 1591.5 lb. (237.5 gal.) in each tank at sta. +181.9 Fuel weight based on 6.70 lb./gal. See NOTE 1 for data on unusable fuel		
Oil Capacity	5.28 gal. total, 3.00 gal. usable (2.3 gal. in each engine-mounted tank at +142.1) See NOTE 1 for data on undrainable oil		
Maximum Operating Altitude	30,000 ft.		
Control Surface Movements	Elevator (horn faired)	Up 14°, +1°, -0°	Down 17°, +1°, -0°
	Elevator trim tabs	Up 8°, +1°, -0°	Down 10°, +2°, -0°
	Rudder (perpendicular to hinge 0° faired with fin)	Right 32°, +1°, -0°	Left 32°, +1°, -0°
	Rudder trim tab (perpendicular to hinge)	Right 11°, +1°, -0°	Left 16°, +1°, -0°
	Aileron	Up 25°, +1°, -0°	Down 14°, +1°, -0°
	Aileron trim tab	Up 19°, +1°, -0°	Down 19°, +1°, -0°
	Wing flap (inboard)		Down 30°, +1°, -0°
	Wing flap (outboard)		Down 20°, +1°, -0°
Serial Nos. Eligible	406-0001 and on		

Data Pertinent to All Models

Datum 100.0 inches forward of forward face of fuselage bulkhead forward of rudder pedals.

Certification Basis **Model 404:**

Part 23 of the Federal Aviation Regulations effective February 1, 1965, as amended by 23-1 through 23-13 except Subpart B as amended through 23-14; and FAR 23.1385(c) as amended through 23-21; and 23.1327 as amended through 23-23. Findings of equivalent level of safety were made for FAR 23.1189(a), 23.1545, and 23.1583(a). In addition, effective S/N 404-0601, FAR 36 dated December 1, 1969, as amended by 36-1 through 36-4.

In addition to the above certification basis, compliance with ice protection has been demonstrated in accordance with FAR 23.1419 of Amendment 23-14 effective December 20, 1973, when ice protection equipment is installed in accordance with Cessna Drawing 5114400, Factory Kit (FK) No. 194, and Pilot's Operating Handbook and FAA Approved Airplane Flight Manual. Aircraft which have been modified in compliance with Accessory Kit (AK) No. 421-106 are considered to be equivalent to those with Factory Kit (FK) No. 194.

Model 406:

Part 23 of the Federal Aviation Regulations effective February 1, 1965, as amended by 23-1 through 23-13 except Subpart B as amended through 23-14; and 23.427, 23.929, 23.979, 23.1017, 23.1027, 23.1163, 23.1182, 23.1189, 23.1192 as amended through 23-14; 23.951, 23.997, 23.1013, 23.1015, 23.1019(a)(1), 23.1019(a)(2), 23.1019(a)(4), 23.1019(a)(5), 23.1019(b), 23.1183 as amended through 23-15; 23.933, 23.971, 23.977, 23.999, 23.1111, 23.1125, 23.1143, 23.1165, 23.1303 (a through d), 23.1385(c), 23.1549 as amended through 23-17; 23.901, 23.939, 23.943, 23.959, 23.967, 23.973, 23.975, 23.995, 23.1093, 23.1121, 23.1141, 23.1145, 23.1193, 23.1203, 23.1305 (a through u and w), 23.1337 as amended through 23-18; 23.1323, 23.1325, 23.1327, 23.1351, 23.1357, 23.1547 as amended through 23-20; 23.45, 23.49, 23.65, 23.67, 23.77, 23.161, 23.1043, 23.1353, 23.1521 as amended through 23-21; 23.1545 as amended through 23-23; 23.903, 23.1529 as amended through 23-26; SFAR 27 as amended by 27-1 through 27-4; Part 36 as amended by 36-1 through 36-12; SFAR 41C; and Exemption No. 4661 from exact compliance with the requirements of Section 23.207(c). Findings of equivalent level of safety were made for FAR 23.1189(a), and that design of

the elevator tab control system provides the level of safety intended by the requirements of FAR 21.21(b)(2) by preventing an unsafe condition. Therefore, FAR 23.629(f), as amended by Amendment 23-23, is applicable to the elevator tab control system, in addition to other requirements in the cited certification basis.

In addition to the above certification basis, compliance with ice protection has been demonstrated in accordance with FAR 23.773 and 23.1419 of Amendment 23-14, FAR 23.1309 as amended through Amendment 23-17, and FAR 23.1416 of Amendment 23-23 when ice protection equipment is installed in accordance with Cessna Drawing 6015006, Factory Kit (FK) No. 194, and Pilot's Operating Handbook and FAA Approved Airplane Flight Manual. Aircraft which have been modified in accordance with Accessory Kit (AK) No. 421-106 are considered to be equivalent to those with Factory Kit (FK) No. 194.

Application for type certificate dated October 9, 1973. Type Certificate No. A25CE issued July 21, 1976, obtained by the manufacturer under delegation option procedures.

Production Basis Production Certificate No. 312 issued and Delegation Option Manufacturer No. CE-3 authorized to issue airworthiness certificates under delegation option provisions of Part 21 of the Federal Aviation Regulations. Effective February 15, 1985, and on, Production Certificate No. 4 is applicable to all spares production for the Model 404 and to all production on the Model 406.

Equipment The basic required equipment as prescribed in the applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification. In addition, the following item of equipment is required:

Stall Warning Indicator - Cessna Dwg. 5818008 (404), 5718030 (406) or
Angle-of-Attack Indicator System - Cessna Dwg. 0800302 (404)

NOTE 1. Current weight and balance report together with list of equipment included in certificated empty weight and loading instructions when necessary must be provided for each aircraft at the time of original certification.

The certified empty weight and corresponding center of gravity location must include undrainable oil (not included in Oil Capacity) and unusable fuel as follows:

(a) S/N 404-0001 through 404-0200

Fuel 48 lb. at (+177.6)

S/N 404-0201 and up

Fuel 28 lb. at (+177.6)

S/N 406-0001 and up

Fuel 44 lb. (6.5 gal.) at (+186.7)

(b) Oil 0.0 lb.

NOTE 2. The placards specified in the FAA Approved Airplane Flight Manual must be displayed.

NOTE 3. (404-0001 and up)

1%, by volume, isopropyl alcohol approved for use as fuel anti-icing additive when used as outlined in Cessna Service Letter ME73-25 dated November 2, 1973, or subsequent revisions.

NOTE 4. (S/N 404-0201 and up), (406-0001 and up)

An optional cargo configuration is available which excludes the passenger air distribution and seating. Such airplanes may be operated with passenger seats installed provided the operating rules for supplemental oxygen are complied with.

NOTE 5. Aircraft operators must observe limitations and placards shown in the applicable Pilot's Operating Manual and FAA Approved Airplane Flight Manual, or later approved revisions as listed below:

Cessna P/N D1540-3-13:	Model 404 Serial 404-0001 through 404-0136 (1977 Model)
Cessna P/N D1563-1-13:	Model 404 Serial 404-0201 through 404-0246 (1978 Model)
Cessna P/N D1572-2-13PH:	Model 404 Serial 404-0401 through 404-0460 (1979 Model)
Cessna P/N D1583-3-13PH:	Model 404 Serial 404-0601 through 404-0695 (1980 Model)
Cessna P/N D1593-1-13PH:	Model 404 Serial 404-0801 through 404-0859 (1981 Model)
Cessna P/N D1624-13PH:	Model 406 Serial 406-0001 and on

NOTE 6. The Model 406 type design has been duplicated as Model F406 in Type Certificate No. A54EU. The type design file is common between Models 406 and F406 and duplicates of the type design file are maintained by the respective type certificate holders.

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