



NOTE: MAXIMUM TAKEOFF rating is the maximum certified for takeoff operation. This rating is referred to as Max Contingency in airplane markings and the Airplane Flight Manual. When the TAKEOFF mode is selected the MAXIMUM TAKEOFF rating is available through actuation of the fuel control Automatic Takeoff Power Control System (ATPCS) when engine is operating at the NORMAL TAKEOFF rating or manually by throttle movement.

NORMAL TAKEOFF rating is the maximum thrust to be set for takeoff operation with the aircraft Automatic Takeoff Power Control System (ATPCS) operative. This rating is referred to as Takeoff in airplane markings and the Airplane Flight Manual. When set, this rating ensures that the MAXIMUM TAKEOFF rating will be achieved upon actuation of ATPCS.

- (a) Overspeed due to propeller system malfunction permissible up to 115% for Max 15 minutes subject to maintenance action.
- (b) Oil temperature must be maintained above 45°C to ensure protection against intake icing.
- (c) Below 56 psi reduce power (NH less than 75%), below 41 psi shutdown.
- (d) Below flight idle.
- (e) 5 seconds
- (f) Power above flight idle not allowed for oil temperature below 0°C.
- (g) Never exceed torque limit (excluding transients) - 103.9% at a maximum NP of 85%.
- (h) The above limitations are defined by the torque capacity of the engine/gearbox combination. The torque settings for normal operation are given in the airplane flight manual.

Fuel:

<u>Grade (Type)</u>	<u>Specification</u>
Kerosene Type	D Eng RD 2453 Issue 4 Am 1
(AVTUR:JP1)	D Eng RD 2494 Issue 8 Am 1
	ASTM D1655-74 Jet A
	ASTM D1655-74 Jet A-1
	3-GP-23h
	IATA Nov 1978: Kerosene Type
Wide Cut Fuels	D Eng RD 2454 Issue 4 Am 1
(AVTAG:JP4)	D Eng RD 2486 Issue 9 Am 1
	ASTM D1655
	IATA Nov 1978 JP 4 type 3-GP-22h
	MIL-T-5624J JP4 grade
	AIR 3407/8 (French)
	TL 9130-006 Issue 4 (German)
	CAN 2-3 22M 78 (Canadian)
High Flash Point	D Eng RD 2498 Issue 7
(AVCAT: JP5)	D Eng RD 2452 Issue 2
	3-GP-24g (Canadian)
	MIL-T-5624J JP5 grade
	AIR 3404/C (French)
	TL 9130-007 Issue 4 (German)
Emergency Fuels	MIL-G-5572 aviation gasoline
	Limited to 150 hours use.
	Engines to be removed for overhaul after this time period.

Fuel Temperature Limits: Aircraft must not be operated with fuel temperature exceeding 54°C or below -48°C.

Propeller and Propeller Limits: Two British Aerospace Hamilton Standard 6/5500/F-1.  
 Diameter 13.75 feet.  
 Pitch settings at 0.75 radius station.  
 Maximum reverse -16°, Ground Idle -7.9°  
 Flight Idle 12.5°  
 Takeoff 10.5°  
 Low pitch warning below 10°, Feathered 76°

Speed range: Takeoff, maximum continuous and maximum takeoff 1,212 rpm.  
 Climb and cruise 1,020 rpm. Flight Range 950 to 1,212 rpm. Quiet taxi mode 825 to 1,212 rpm.  
 Any ground check requiring continuous operation above 920 prop rpm or between 620 and 720 prop rpm is prohibited except for pre-flight checks of short duration.

## Airspeed Limits:

$V_{MO}$  (Maximum Operating)  
 227 knots IAS at sea level increasing linearly to  
 230 knots IAS up to 15,000 ft.  
 220 knots IAS above 15,000 ft increasing linearly to  
 221 knots IAS up to 22,000 ft decreasing linearly to  
 208 knots IAS at 25,000 ft.

$V_A$  (Maneuvering - flaps retracted)  
 170 knots IAS at all altitudes

$V_{FE}$  (Flap Speeds)  
 180 knots IAS 7° flap  
 180 knots IAS 15° flap  
 150 knots IAS 20° flap (with Mod 05000A)  
 140 knots IAS 22° flap  
 130 knots IAS 27° flap  
 140 knots IAS 29° flap (with Mod 05000A)

$V_{LO}$  (Landing Gear Operation)  
 180 knots IAS Lowering  
 150 knots IAS Retraction

$V_{LE}$  (Landing Gear Extended)  
 180 knots IAS

 $V_{MC}$  Minimum Control Speeds:

In the air, $V_{MC}$ =	97 knots IAS	0° or 7° flap
	92 knots IAS	15° flap
On or near the ground $V_{mcg}$ =	99 knots IAS	7° flap up to 38,500 lbs increasing linearly to
	101.5 knots IAS	at 50,500 lbs
With 15° flap $V_{mcg}$ =	95 knots IAS	at 38,500 lbs. increasing linearly to
	100 knots IAS	at 50,550 lbs.
During landing $V_{mcl}$ =	90 knots IAS	with 20° flap (with Mod 05000A)
	90 knots IAS	with 22° flap
	88 knots IAS	with 27° flap
	88 knots IAS	with 29° flap (with Mod 05000A)

CG Range (Landing Gear Extended):

Landing Gear Retraction Moment - 81,719 lb. in.

Weight (lb)	Forward (In Flight)	Forward (Takeoff & Landing)	AFT
	Aft of Datum (ins)	Aft of Datum (ins)	Aft of Datum (ins)
28,000	49.0	51.4	66.5
30,000	-	-	66.5
33,070	49.0	-	-
33,400	-	51.4	-
38,000	-	-	75.5
46,800	60.05	62.2	76.85
50,550	61.7	63.7	77.25

Straight line variations between weights.

NOTE: With 27° or 29° flap selected for landing, the forward CG limit is 58.0 inches aft of Datum below 41,430 lbs.

Maximum Weights:      Maximum Ramp Weight                      50,700 lbs  
 Maximum Takeoff Weight    50,550 lbs  
 Maximum Landing Weight    49,050 lbs  
 Maximum Zero Fuel Weight 46,800 lbs

Maximum Baggage:      Baggage hold located according to passenger layout.

Between fuselage formers 378 in. forward and 252 in. forward of the fuselage datum, maximum permissible floor loading is 100 lb/ft<sup>2</sup> over the full width of the fuselage and a maximum load of 1,700 lbs.

Between fuselage formers 308 in. aft and 378 in. aft of the fuselage datum, maximum permissible floor loading is 100 lb/ft<sup>2</sup> over the full width of the fuselage and a maximum load of 2,325 lbs.

Flying Control Surface Angular Travel:

Rudder	19° + 1° - 0°	Right	19° + 1° - 0°	Left
Spring Tab (rudder locked)	20° + 3° - 1.5°	Right	20° + 3° - 1.5°	Left
Gearred Trim Tab (trim movement)	9.15° + 1° - 0°	Right	15.15° + 1° - 0°	Left
Ailerons	20° ± 0.5°	Up	20° + 0.5°	Down
Trim Tabs (right aileron only)	15.5° + 1.5° - 0°	Up	15.5° + 1.5° - 0°	Down
Gearred Tabs	5°	Up	1° to 3.5°	Down
Elevator	22° + 1° - 0°	Up	13.5° + 1° - 0°	Down
Tab (Balance mode)	5.0° ± 1°	Up	8.1° ± 1°	Down
Tab (trim mode)	3.2° ± 1°	Up	11.2° ± 1°	Down

Flaps total angle of travel 27° or 29° (with Mod 05000A) The rigging instructions including tolerances are given in the Manufacturers Recommended Maintenance Manual.

Minimum Crew:                      2 - Pilot and Co-pilot

Maximum Passengers:              68

Maximum Operating Altitude:    25,000 ft

Fuel Capacity:                      Two integral wing fuel tanks each of 854 US usable gallons capacity. Moment arm plus 82.3 in. (i.e. aft of CG datum point).  
 Maximum fuel density 7.1 lb/U.S. gallon  
 Minimum fuel density 6.0 lb./U.S. gallon

Oil Capacity Movement:	Two oil tanks each 5.72 US gallons capacity, including propeller pitch control oil. arm - 10 in. (i.e. forward of CG datum point).
Datum:	The zero datum of the airplane is the forward face of the rear spar former. The CG datum is 108.3 in. forward of zero datum.
Standard Mean Chord (SMC):	The standard mean chord is 100.632 in. The leading edge of the standard mean chord is 44 in. aft of the CG datum.
Leveling Means:	The seat rail in the forward fuselage adjacent to the left hand forward passenger door.
Operating Limitations:	Aircraft shall be operated in compliance with the operating limitations specified in the CAA approved Flight Manual Document No ATP 002 dated August 25, 1988, or for aircraft with Mod 05000A Flight Manual Document No. ATP 0004 dated December 14, 1989 or later approved revisions.
Serial Nos. Eligible:	A United Kingdom Certificate of Airworthiness for Export endorsed as noted under "Import Requirements" below must be submitted for each individual airplane for which application for United States certification is made.
Transition Statement:	<i>The United Kingdom Civil Aviation Authority originally type certificated this aircraft under its type certificate Number BA23. The FAA validated this product under U.S. Type Certificate Number A24EU. Effective September 28, 2003, the European Aviation Safety Agency (EASA) began oversight of this product on behalf The United Kingdom.</i>
Import requirement	The FAA can issue a U.S. airworthiness certificate based on an NAA Export Certificate of Airworthiness (Export C of A) signed by a representative of the United Kingdom Civil Aviation Authority on behalf of the European Community. The Export C of A should contain the following statement: 'The aircraft covered by this certificate has been examined, tested, and found to conform with Type Design approved under U.S. Type Certificate No. A24EU and to be in a condition for safe operation.'
Service information	<p>Each of the documents listed below that contain a statement that it is approved by the European Aviation Safety Agency (EASA) - or for approvals made before September 28, 2003 - by the United Kingdom Civil Aviation Authority, are accepted by the FAA and are considered FAA approved. <u>Additionally, approvals issued by BAE Systems (Operations) Limited under the authority of EASA approved Design Organization EASA.21J.047 - or for approvals made before September 28, 2003 - under the authority of United Kingdom Civil Aviation Authority or Design Organization Approval No. CAA.JA.02034 are considered FAA approved. These approvals pertain to the type design only.</u></p> <ul style="list-style-type: none"> <li>•BAE Systems (Operations) Limited Service Bulletins, except as noted below,</li> <li>• Structural repair manuals,</li> <li>• Vendor manuals referenced BAE Systems (Operations) Limited service bulletins</li> <li>• Aircraft flight manuals,</li> <li>• Repair Instructions.</li> </ul>

Note: Design changes that are contained in BAE Systems (Operations) Limited Service Bulletins and that are classified as Level 1 Major in accordance with either the US United Kingdom Civil Aviation Authority or US/EASA Bilateral Aviation Safety Agreement Implementation Procedures for Airworthiness must be approved by the FAA.

- Certification Basis:
1. JAR 25 including changes 1 through 10
  2. The following section of FAR Part 25 effective February 1, 1965 as amended through Amendment 25-56:

25.1041	25.671(c)(1)	25.954	25.1305 (c)(6) and (7)
25.1091	25.693	25.993 (c)	25.1309 (a), (b), (c), (d), (e)
25.1093 (b)(1)(i) and (ii), (b)(2)	25.729 (e)	25.994	25.1331 (a)(3)
25.1103	25.735 (f)(1)	25.1013	25.1337 (a)(1) and (3)
25.119 (b)	25.773 (b)(1)(ii), (b)(2)	25.1015	25.1353 (a), (c)(5), (c)(6)
25.125 (a)	25.777 (e), (g)	25.1019	25.1401 (b), (f)
25.201(d)	25.781	25.1041	25.1411 (a)(2)
25.331 (c)(1)	25.783(f)	25.1091	25.1415 (d), (e)
25.351 (a)(1)	25.785 (g)	25.1093 (b)(1)(i) and (ii), (b)(2)	25.1438 (a),(b), (c)
25.361 (b)	25.787 (a)	25.1103	25.1501
25.335 (b)(2)	25.803	25.1121	25.1513
25.365 (a), (d)	25.809(j)	25.1123	25.1521 (b) and (c)
25.511 (b)(6)	25.832	25.1141 (f)(2)	25.1522
25.571 (b)(6)	25.843 (a)	25.1142	25.1547 (c)
25.581	25.901 (c)	25.1143 (f)	25.1549
25.613	25.903 (d)(1)	25.1181	25.1583 (a) (4) , (I)
25.615	25.939 (a)	25.1191	25.1585 (a)(1) through (10)
25.621	25.943	25.1195	25.1587 (a), (b) (4)
25.631	25.951 (a)	25.1303 (c)(1)	

3. The applicant elected to have the following later requirements added to the TC basis:
  - 25.812 as amended by Amendment 25-58
  - 25.853 as amended by Amendment 25-59
  - 25.904 as amended by Amendment 25-62
4. Special Condition 25-ANM-17 dated March 9, 1988, concerning engine control lightning protection.
5. Exemption No. 4812 docket No 013NM concerning 25.571 (e)(2) sustaining structural damage from the loss of a propeller blade impact.
6. Equivalent safety findings exist with respect to the following sections of the FAR:
  - Section 25.783 (f) via pressurization test.
  - Section 25.773 (b)(2) by additional design requirements and flight evaluation.
7. Special Federal Aviation Regulation No 27, effective February 1, 1974, including Amendments 27-1 through 27-6 (Fuel Venting and exhaust Emissions).
8. Part 36, effective December 1, 1969, including Amendment 36-1 through 36-14. Amendment 36-1 through 36-16 (Mod 050004).
9. BAE elected to demonstrate compliance with 25.1419: Ice Protection. Date of Application for Type Certificate April 5, 1982

In accordance with 14 CFR Section 21.29, the, FAA has determined the above noted certification basis item 1 through 6 are equivalent to or exceed FAR Part 25 as amended through 25-56 requirements.

Required Equipment: The basic required equipment as prescribed in the applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification.

Federal Aviation Administration (FAA) approved Airplane Flight Manual, BAE Model ATP, Document No ATP 002 dated August 25, 1988, published in the English language (CAA-UK approved on behalf of the FAA) or later CAA-UK approved revision, is required.

For aircraft with Modification 05000A Federal Aviation Administration (FAA) approved Airplane Flight Manual, Bae Model ATP, Document No ATP 004 dated December 14, 1989 published in the English language (CAA approved on behalf of the FAA) or later CAA-UK approved revision, is required.

Airplane Weight and Balance Manual Number BAe-MWE-R-ATP-2000 series.

Available Documents: Maintenance Review Board Report, Document No MRB ATP-01

NOTE 1 Weight and Balance:

- (a) Current Weight and Balance Manual included in certificated empty weight and loading instructions must be provided for each aircraft at the time of original certification and at all times subsequently. British Aerospace "Weight and Balance Manual" No BAe-MWE-R-ATP-2000 series contains all the loading information required for each aircraft in its delivery configuration.
- (b) The airplane must be loaded so that the CG is within the specified limits at all times when all influences on CG position have been considered.
- (c) The Weight and Balance defined in (a) above quotes the quantities of usable fuel as determined by the critical conditions of FAR 25.959. Undrainable fuel, being that remaining in the aircraft when all fuel drain valves are opened on the ground, is also quoted in the Weight and Balance Manual.

NOTE 2 All placards required in the Approved Airplane Flight Manual must be installed in the appropriate location.

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