

United States Of America  
Department of Transportation - Federal Aviation Administration  
**Supplemental Type Certificate**

*Number* SA02357LA

*This Certificate issued to* Blackhawk Modifications Inc.  
7601 Karl May Drive  
Waco, Texas 76708

*certifies that the change in the type design for the following product with the limitations and conditions therefor as specified hereon meets the airworthiness requirements of Part 23\* of the Federal Aviation Regulations. \*Certification basis set forth on continuation sheet 2*

*Original Product Type Certificate Number:* A37CE

*Make:* Cessna

*Model:* 208B

*Description of Type Design Change:* Replacement of the original Pratt and Whitney PT6A-114(A) engine and propeller with a Pratt and Whitney PT6A-42A and 4-blade Hartzell Propeller HC-E4N-3P/D9900(K) propeller or 3-blade Hartzell Propeller HC-B3TN-3AE(Y)/T10290N(K)+2, and associated equipment in accordance with FAA Approved Master Drawing List 200803-000, Rev IR, dated June 8, 2011, or later FAA approved revisions. Pilot's Operating Handbook and FAA Approved Airplane Flight Manual Supplement, AFMS 200803, Rev IR, dated June 22, 2011 or later FAA approved revisions. Blackhawk Modifications Inc. Instructions for Continued Airworthiness document ICA 200803-30, Rev A, dated May 2011, or later revisions. Installation of optional Howell Instruments, Inc. gages as acceptable alternates for the existing Ultra Electronic FlightLine Systems gages in accordance with FAA Approved Master Drawing List 201307-000, Rev IR, dated Sep 12, 2014, or later FAA approved revisions and Pilot's Operating Handbook and FAA Approved Airplane Flight Manual Supplement, AFMS 201307, Rev IR, dated January 16, 2015, or later FAA approved revisions.

*Limitations and Conditions:* The installation should not be incorporated in any aircraft unless it is determined that the interrelationship between this installation and any previously approved configuration will not introduce any adverse effect upon the airworthiness of that aircraft. The approval of this modification applies to the above-noted airplane models only. If the holder agrees to permit another person to use this certificate to alter the product, the holder shall give the other person written evidence of that permission.

(CONTINUED on page 3)

*This certificate and the supporting data which is the basis for approval shall remain in effect until surrendered, suspended, revoked or a termination date is otherwise established by the Administrator of the Federal Aviation Administration.*

*Date of application:* April 28, 2009

*Date reissued:*

*Date of issuance:* June 23, 2011

*Date amended:* April 16, 2012, October 25, 2012, November 14, 2013, January 28, 2015



*By direction of the Administrator*

(Signature)

Manager, Propulsion Branch  
Los Angeles Aircraft Certification Office

(Title)

Any alteration of this certificate is punishable by a fine of not exceeding \$1,000, or imprisonment not exceeding 3 years, or both.

# Supplemental Type Certificate

## (Continuation Sheet)

### *Limitations and Conditions (continued)*

- AeroAcoustics Aircraft Systems, Inc., "Aircraft Payload Extender STOL System," STC SA01805SE is required as part of this alteration.
- This modification is not compatible with Cessna Model 208B Serial Number 208B1190, 208B1216, and 208B2000 and greater (Garmin 1000 installations) and aircraft equipped with STC SA01478WI or Cessna SK208-174 (TKS installation).

### *Certification Basis*

The certification basis for this modification is as follows:

14 CFR part 23 of the Federal Aviation Regulations effective February 1, 1965, as amended by Amendments 23-1 through 23-59 for 23.1, 23.21, 23.23, 23.25, 23.33, 23.45, 23.49, 23.51, 23.53, 23.63, 23.65, 23.69, 23.73, 23.75, 23.77, 23.141, 23.143, 23.145, 23.147, 23.153, 23.155, 23.157, 23.161, 23.171, 23.173, 23.175, 23.177, 23.179, 23.181, 23.201, 23.203, 23.221, 23.231, 23.233, 23.251, 23.253, 23.301, 23.303, 23.305, 23.307, 23.321, 23.331, 23.333, 23.335, 23.337, 23.341, 23.361, 23.363, 23.371, 23.421, 23.471, 23.473, 23.479, 23.481, 23.483, 23.485, 23.493, 23.499, 23.507, 23.509, 23.561, 23.572, 23.601, 23.603, 23.605, 23.607, 23.609, 23.611, 23.613, 23.619, 23.623, 23.625, 23.627, 23.629, 23.735, 23.771, 23.831, 23.863, 23.865, 23.867, 23.901, 23.903, 23.905, 23.907, 23.925, 23.929, 23.933, 23.939, 23.955, 23.959, 23.961, 23.991, 23.993, 23.994, 23.999, 23.1011, 23.1017, 23.1019, 23.1021, 23.1023, 23.1027, 23.1041, 23.1043, 23.1045, 23.1091, 23.1093, 23.1103, 23.1111, 23.1121, 23.1123, 23.1141, 23.1163, 23.1165, 23.1181, 23.1183, 23.1193, 23.1203, 23.1301, 23.1305, 23.1308, 23.1309, 23.1321, 23.1322, 23.1327, 23.1329, 23.1331, 23.1337, 23.1351, 23.1357, 23.1359, 23.1365, 23.1381, 23.1416, 23.1431, 23.1501, 23.1507, 23.1519, 23.1521, 23.1525, 23.1527, 23.1529, 23.1541, 23.1543, 23.1547, 23.1549, 23.1551, 23.1555, 23.1557, 23.1581, 23.1583, 23.1585, 23.1587, 23.1589, except 23.1419 as amended by 23-1 through 23-14.

1. Special Condition 23-ACE-3, Dynamic Evaluation, Engine Installation. Factors from this special condition were incorporated into the whirl mode flutter analysis per 14 CFR 23.629(e)(1) and (2) Amendment 23-54, Flutter.

2. 14 CFR 34, effective January 1, 1994, as amended by Amendments 34-1 thru 34-3.

3. 14 CFR 36, effective December 1, 1969, as amended by Amendments 36-1 thru 36-28.

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