

United States Of America
Department of Transportation - Federal Aviation Administration

Supplemental Type Certificate

Number SA00323BO

This Certificate issued to Avidyne Corporation
55 Old Bedford Road
Lincoln, Massachusetts 01773

certifies that the change in the type design for the following product with the limitations and conditions therefor as specified hereon meets the airworthiness requirements of Part 23 of the Federal Aviation Regulations.

Original Product Type Certificate Number: 3A13
Make: Cessna Aircraft Company
Model: 182E, 182F 182G, 182H, 182J, 182K, 182L, 182M, 182N, 182P, 182Q, 182R, 182S, 182T, R182

Description of Type Design Change:

Installation of Avidyne Corporation Autopilot:

- Model DFC90, p/n 700-00170-022 (28-volt electrical system) or p/n 700-00170-024 (14-volt electrical system), loaded with Release 0 software (or later FAA approved release)

as a replacement for the S-TEC Autopilot (see Limitations and Conditions number 3) in accordance with the Avidyne Corporation Master Document List, Document AVDFC-115, Rev 00, dated September 25, 2012, or later FAA-approved revisions.

Limitations and Conditions:

1. Operation must be in accordance with Avidyne Corporation Aircraft Flight Manual Supplement (AFMS) Document 600-00293-000, Rev 00, FAA approved on October 01, 2012, or later FAA-approved revision.

The AFMS must be carried in the aircraft during all flights.

(Limitations and Conditions continued on Page 3 of 3)

This certificate and the supporting data which is the basis for approval shall remain in effect until surrendered, suspended, revoked or a termination date is otherwise established by the Administrator of the Federal Aviation Administration.

Date of application: March 02, 2012

Date reissued:

Date of issuance: October 01, 2012

Date amended:

By direction of the Administrator



(Signature)

Robert G. Mann
Manager
Boston Aircraft Certification Office

(Title)

United States Of America
Department of Transportation - Federal Aviation Administration

Supplemental Type Certificate

(Continuation Sheet)

Number SA00323BO

October 01, 2012

INSTRUCTIONS: The transfer endorsement below may be used to notify the appropriate FAA Regional Office of the transfer of the Supplemental Type Certificate.

The FAA will reissue the certificate in the name of the transferee and forward it to him.

TRANSFER ENDORSEMENT

Transfer the ownership of Supplemental Type Certificate Number _____

to (*Name of transferee*) _____

(*Address of transferee*) _____

(*Number and street*)

(*City, State, and ZIP code*)

from (*Name of grantor*)(*Print or type*) _____

(*Address of grantor*) _____

(*Number & street*)

(*City, State, and ZIP code*)

Extent of Authority (if licensing agreement): _____

Date of Transfer: _____

Signature of grantor (*In ink*): _____

United States Of America
Department of Transportation - Federal Aviation Administration

Supplemental Type Certificate

(Continuation Sheet)

Number SA00323BO

October 01, 2012

Limitations and Conditions: (continued from Page 1 of 3)

2. Maintenance must be in accordance with Instructions for Continued Airworthiness (ICA), Document AVDFC-125, Rev 00, dated August 09, 2012, or later FAA accepted revision.

The ICA must be made available to the operator at the time of installation.

3. Aircraft must be configured with an existing S-TEC Manual Electric Trim (MET) System and with an existing S-TEC Autopilot as follows in order to accomplish this STC:

Existing S-TEC Autopilot	S-TEC Part Number
System 20 with 30ALT	01260(-), 01261(-)
System 30	1260(-), 1261(-)
System 40 with 30ALT	0129(-), 0130(-), 01261(-)
System 50	0131(-), 0132(-)
System 55(x)	01192(-)
System 60-1, 60-2, 65	0109(-), 0110(-)

4. Aircraft must be configured with an existing Aspen Avionics, Inc. EFD1000 Primary Flight Display, p/n 910-00001-XXX, loaded with Release 2.6 software (or later FAA approved release), in order to accomplish this STC.
5. Maximum aft center of gravity is 46.0 inches aft of reference datum.
6. Compatibility of this design with previously approved modifications must be determined by the installer.

Certification Basis:

Based on 14 CFR § 21.115 and § 21.101, and the FAA policy for design changes that are identified as not significant in FAA Order 8110.48, the certification basis for the Cessna Aircraft Company 182E, 182F 182G, 182H, 182J, 182K, 182L, 182M, 182N, 182P, 182Q, 182R, 182S, 182T, R182 aircraft is as follows:

- a. The type certification basis for parts **not changed or not affected** by the change is shown on TCDS 3A13.
- b. The modification certification basis for parts **changed or affected** by the change since the reference application date, March 02, 2012, is based upon part 23 as amended by Amendment 23-62 as follows:

Regulations at the latest amendment 23-0 through 23-62

23.25(a), 23.301, 23.303, 23.305, 23.307(a), 23.337(a)(1)(b)(1), 23.473(d)(g), 23.561(a)(b)(3)(e), 23.601 - 23.613, 23.771(a), 23.773(a)(2), 23.777(a)(b), 23.867(b), 23.1301, 23.1306, 23.1308(b), 23.1309, 23.1311(a)(2)(3)(4), 23.1322, 23.1329, 23.1335, 23.1351(a)(1)(2)(i)(b)(1)(e), 23.1357(a)(2)(b)(c)(d), 23.1359(c), 23.1365(a)(b)(d)(e)(f), 23.1367, 23.1381(a)(b), 23.1431(a)(b)(e), 23.1501, 23.1523, 23.1525, 23.1529, 23.1541(a), 23.1555(a)(b), 23.1581(a)(b)(2)(3)(f), 23.1583(m), 23.1585(j)

Regulations at an intermediate amendment

23.785(e) at 23-34

Regulations at the amendment level in TCDS 3A13

None

If the holder agrees to permit another person to use this certificate to alter the product, the holder shall give the other person written evidence of that permission.

-----END-----