

**FEDERAL AVIATION ADMINISTRATION
AIRWORTHINESS DIRECTIVES**

**SMALL AIRPLANES, ROTORCRAFT, GLIDERS,
BALLOONS, & AIRSHIPS**

BIWEEKLY 2016-13

6/13/2016 - 6/26/2016



Federal Aviation Administration
Continued Operational Safety Policy Section, AIR-141
P.O. Box 25082
Oklahoma City, OK 73125-0460

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SMALL AIRCRAFT, ROTORCRAFT, GLIDERS, BALLOONS, & AIRSHIPS

AD No.	Information	Manufacturer	Applicability
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Information Key: E - Emergency; COR - Correction; S – Supersedes; R - Replaces

Biweekly 2016-01

2015-26-04	S 2002-13-11	Airbus Helicopters	EC120B helicopters
2015-26-08		Piper Aircraft, Inc.	PA-44-180, PA-44-180T airplanes
2015-26-10		Sikorsky Aircraft Corporation	S-76A, S-76B, and S-76C helicopters

Biweekly 2016-02

2015-12-09 R1	R 2015-12-09	Airbus Helicopters Deutschland GmbH	EC135P1, EC135T1, EC135P2, EC135T2, EC135P2+, EC135T2+, and MBB-BK 117 C-2
2016-01-01		Piper Aircraft, Inc.	PA-46-500TP
2016-01-06		Agusta S.p.A.	AB139 and AW139
2016-01-14		Airbus Helicopters Deutschland GmbH	MBB-BK 117 A-1, A-3, A-4, B-1, B-2, C-1, and C-2
2016-01-15		Agusta S.p.A.	AB139 and AW139
2016-01-19		MD Helicopters Inc.	500N and 600N

Biweekly 2016-03

2015-22-51		Agusta S.p.A.	A109A and A109AII helicopters
2016-02-06		Bell Helicopter Textron Canada Limited	429 helicopters

Biweekly 2016-04

2016-03-02		Turbomeca S.A.	ARRIEL 2C, 2C1, 2C2, 2S1, and 2S2 turboshaft engines
2016-03-05	S 2014-13-01	Airbus Helicopters Deutschland GmbH	MBB-BK 117 C-2 and MBB-BK 117 D-2 helicopters
2016-04-05	S 2014-03-18	B-N Group Ltd.	BN-2, BN-2A, BN-2A-2, BN-2A-3, BN-2A-6, BN-2A-8, BN-2A-9, BN-2A-20, BN-2A-21, BN-2A-26, BN-2A-27, BN-2B-20, BN-2B-21, BN-2B-26, BN-2B-27, BN2A MK. III, BN2A MK. III-2, and BN2A MK. III-3 airplanes

Biweekly 2016-05

2016-04-04		M7 Aerospace LLC	SA26-AT, SA226-T(B), SA226-AT, SA226-T, SA226-TC, SA227-AC (C-26A), SA227-AT, SA227-BC (C-26A), SA227-CC, SA227-DC (C-26B), and SA227-TT
2016-04-14		Turbomeca S.A.	Arriel 1E2
2016-04-15		MD Helicopters Inc.	369A, 369D, 369E, 369FF, 369HE, 369HM, 369HS, 500N, and 600N
2016-05-06	S 2014-07-52	Airbus Helicopters	AS350B, AS350BA, AS350B1, AS350B2, AS350B3, AS350C, AS350D, AS350D1, AS355E, AS355F, AS355F1, AS355F2, AS355N, and AS355NP

Biweekly 2016-06

2016-04-12		Turbomeca S.A.	Arriel 2B, 2B1, 2C, 2C1, 2C2, 2D, 2E, 2S1, and 2S2 turboshaft engines
2016-05-01	R 96-12-12	Piper Aircraft, Inc.	PA-31, PA-31-300, PA-31-325 and PA-31-350
2016-05-08	R 2006-23-17	Turbomeca S.A.	Turmo IV A and IV C turboshaft engines.
2016-05-09		MD Helicopters, Inc.	369A (Army OH-6A), 369H, 369HE, 369HM, 369HS, and 369D; 369E, 369F and 369FF, 500N
2016-05-10		Airbus Helicopters	AS 365 N3, EC 155B, and EC155B1
2016-05-11		Sikorsky Aircraft Corporation	S-92A
2016-05-13		Pratt & Whitney Canada Corp.	PT6A-60AG, BS919 and BS1048; PT6A-65AG, BS708, BS903, BS1101, and BS1102; PT6A-67AF; and PT6A-67AG
2016-06-01	S 2007-06-06	B-N Group Ltd.	BN-2, BN-2A, BN-2A-2, BN-2A-3, BN-2A-6, BN-2A-8, BN-2A-9, BN-2A-20, BN-2A-21, BN-2A-26, BN-2A-27, BN-2B-20, BN-2B-21, BN-2B-26, BN-2B-27, BN2A MK. III, BN2A MK. III-2, BN2A MK. III-3 BN2A, BN2B, and BN2A MKIII, BN2A, BN2B, and BN2A MKIII

Biweekly 2016-07

2016-06-09		Turbomeca S.A.	Makila 2A and 2A1
2016-07-01	S 2014-07-04R1	Sikorsky Aircraft Corporation	S-92A
2016-07-02		Honeywell International Inc.	TFE731-4, -4R, -5AR, -5BR, and -5R
2016-07-11		Weatherly Aircraft Company	201, 201A, 201B, 201C, 620, 620A, 620B, 620B-TG, and 620TP

SMALL AIRCRAFT, ROTORCRAFT, GLIDERS, BALLOONS, & AIRSHIPS

AD No.	Information	Manufacturer	Applicability
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Information Key: E - Emergency; COR - Correction; S – Supersedes; R - Replaces

Biweekly 2016-08

2016-07-13		GE Aviation Czech s.r.o	M601E-11
2016-07-19		Technify Motors GmbH	TAE 125-02-99 and TAE 125-02-114
2016-07-21	R 2015-20-13	Piper Aircraft, Inc.	PA-28-161, PA-28-181, and PA-28R-201
2016-07-24		Textron Aviation, Inc.	310 through 310R, E310H, E310J, T310P through T310R, 310J-1, 320 through 320F, 320-1, 335, 340, 340A, 401 through 401B, 402 through 402C, 411, 411A, 414, 414A, and 421 through 421C
2016-07-26	R 2010-23-02	Airbus Helicopters	SA-365N, SA-365N1, AS-365N2, and AS 365 N3
2016-07-27		Airbus Helicopters	SA341G and SA342J
2016-07-29		Airbus Helicopters	EC225LP, AS332C, AS332L, AS332L1, and AS332L2
2016-08-08	S 92-06-10	SOCATA	MS 880B, MS 885, MS 892A-150, MS 892E-150, MS 893A, MS 893E, MS 894A, MS 894E, Rallye 100S, Rallye 150ST, Rallye 150T, Rallye 235E, and Rallye 235C

Biweekly 2016-09

2016-08-16		Turbomeca S.A.	Arriel 2E turboshaft engines
2016-08-17	2010-19-51	Bell Helicopter Textron Canada	222, 222B, 222U, 230, and 430 helicopters
2016-08-21		Kaman Aerospace Corporation	K-1200 helicopters

Biweekly 2016-10

2015-09-04 R1	R 2015-09-04	DG Flugzeugbau GmbH	DG-1000T gliders
2016-06-06		Quest Aircraft Design, LLC	KODIAK 100 airplanes
2016-08-18		Piper Aircraft, Inc	PA-31-350 airplanes
2016-08-19		Mitsubishi Heavy Industries, Ltd	MU-2B-30, MU-2B-35, and MU-2B-36 , MU-2B-36A and MU-2B-60 airplanes,
2016-08-20	S 2014-12-51	Airbus Helicopters (Previously Eurocopter France)	EC130B4 and EC130T2
2016-09-02		Turbomeca S.A.	Astazou XIV B and XIV H turboshaft engines
2016-09-09	S 2013-08-17	Airbus Helicopters (Previously Eurocopter France)	SA-365N, SA-365N1, AS-365N2, AS 365 N3, and SA-366G1 helicopters
2016-10-01		M7 Aerospace LLC	SA226-AT, SA226-T, SA226-T (B), SA226-TC, SA227-AC (C-26A), SA227-AT, SA227-BC (C-26A), SA227-CC, SA227-DC (C-26B), and SA227-TT airplanes
2016-10-03		Viking Air Limited	DHC-3 airplanes

Biweekly 2016-11

2016-10-03	COR.	Viking Air Limited	DHC-3 airplanes
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Biweekly 2016-12

2016-11-09		Turbomeca S.A.	Arriel 1D and 1D1
2016-11-10	S 2000-20-11	BLANIK LIMITED	L-13 Blanik and L-13 AC Blanik
2016-11-11		EVEKTOR, spol. s.r.o.	L 13 SEH VIVAT and L 13 SDM VIVAT
2016-11-12	S 2000-20-12	EVEKTOR, spol. s.r.o.	L 13 SEH VIVAT and L 13 SDM VIVAT
2016-11-13	S 99-19-33	BLANIK LIMITED	L-13 Blanik and L-13 AC Blanik
2016-11-20		B/E Aerospace	Protective Breathing Equipment (PBE)
2016-11-21		Airbus Helicopters Deutschland GmbH	EC135P1, EC135P2, EC135P2+, EC135T1, EC135T2, and EC135T2+
2016-12-01		Pilatus Aircraft LTD.	PC-12, PC-12/45, PC-12/47, and PC-12/47E
2016-12-02		Various Aircraft	See AD
2016-12-51	E	Airbus Helicopters	AS332L2 and Model EC225LP

Biweekly 2016-13

2016-12-06		Turbomeca S.A.	MAKILA 2A and MAKILA 2A1 turboshaft engines
2016-12-07	S 2010-11-10	Turbomeca S.A.	Astazou XIV B and XIV H turboshaft engines
2016-12-08		GROB Aircraft AG	G115EG airplanes
2016-12-13	S 2000-05-17 S 2001-04-12	Airbus Helicopters	EC120B helicopters
2016-13-04		BRP-Powertrain GmbH & Co KG	Rotax model 912 F2, 912 F3, 912 F4, 912 S2, 912 S3, 912 S4, 914 F2, 914 F3, and 914 F4 reciprocating engines



2016-12-06 Turbomeca S.A.: Amendment 39-18555; Docket No. FAA-2015-8257; Directorate Identifier 2015-NE-36-AD.

(a) Effective Date

This AD becomes effective July 22, 2016.

(b) Affected ADs

None.

(c) Applicability

This AD applies to Turbomeca S.A. MAKILA 2A and MAKILA 2A1 turboshaft engine models with a high-pressure gas generator module (M03) that has modification (mod) TU 52 installed.

(d) Reason

This AD was prompted by two occurrences of crack initiation on a ferrule of the diffuser, which propagated and led to the ferrule rupture. We are issuing this AD to prevent rupture of the ferrule of the diffuser, which could result in engine fire and damage to the helicopter.

(e) Actions and Compliance

Comply with this AD within the compliance times specified, unless already done.

(1) Borescope inspect the centrifugal diffuser ferrule, part number 0298210100, prior to the ferrule accumulating 700 hours, time since new or time since replacement or within 30 hours from the effective date of this AD, whichever is later. Use Accomplishment Instructions, paragraphs 2.4.1 through 2.4.2.2.1, of Turbomeca S.A. Alert Mandatory Service Bulletin (MSB) No. 298 72 2832, Version C, dated April 15, 2016, to do the borescope inspections required by this AD.

(2) Repeat the borescope inspection required by this AD every 300 hours since last inspection.

(3) If any crack, loss of contact between the ferrule and diffuser axial vane, or any contact between the injection manifold supply pipe and the diffuser ferrule is found, remove the diffuser case and replace the ferrule with a part eligible for installation.

(f) Credit for Previous Actions

You may take credit for the actions required by paragraph (e) of this AD if you performed those actions using Turbomeca S.A. MSB No. 298 72 2832, Version B, dated October 12, 2015 or earlier versions, before the effective date of this AD.

(g) Alternative Methods of Compliance (AMOCs)

The Manager, Engine Certification Office, FAA, may approve AMOCs for this AD. Use the procedures found in 14 CFR 39.19 to make your request. You may email your request to: ANE-AD-AMOC@faa.gov.

(h) Related Information

(1) For more information about this AD, contact Brian Kierstead, Aerospace Engineer, Engine Certification Office, FAA, Engine & Propeller Directorate, 1200 District Avenue, Burlington, MA 01803; phone: 781-238-7772; fax: 781-238-7199; email: brian.kierstead@faa.gov.

(2) Refer to MCAI European Aviation Safety Agency AD 2015-0209R1, dated April 20, 2016, for more information. You may examine the MCAI in the AD docket on the Internet at <http://www.regulations.gov> by searching for and locating it in Docket No. FAA-2015-8257.

(i) Material Incorporated by Reference

(1) The Director of the Federal Register approved the incorporation by reference (IBR) of the service information listed in this paragraph under 5 U.S.C. 552(a) and 1 CFR part 51.

(2) You must use this service information as applicable to do the actions required by this AD, unless the AD specifies otherwise.

(i) Turbomeca S.A. Alert MSB No. A298 72 2832, Version C, dated April 15, 2016.

(ii) Reserved.

(3) For Turbomeca S.A. service information identified in this AD, contact Turbomeca S.A., 40220 Tarnos, France; phone: (33) 05 59 74 40 00; fax: (33) 05 59 74 45 15.

(4) You may view this service information at FAA, Engine & Propeller Directorate, 1200 District Avenue, Burlington, MA. For information on the availability of this material at the FAA, call 781-238-7125.

(5) You may view this service information at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call 202-741-6030, or go to: <http://www.archives.gov/federal-register/cfr/ibr-locations.html>.

Issued in Burlington, Massachusetts, on June 10, 2016.
Colleen M. D'Alessandro,
Manager, Engine & Propeller Directorate,
Aircraft Certification Service.



2016-12-07 Turbomeca S.A.: Amendment 39-18556; Docket No. FAA-2010-0219; Directorate Identifier 2010-NE-14-AD.

(a) Effective Date

This AD is effective July 26, 2016.

(b) Affected ADs

This AD supersedes AD 2010-11-10.

(c) Applicability

This AD applies to Turbomeca S.A., Astazou XIV B and XIV H turboshaft engines with the following part number (P/N) and serial number (S/N) third stage turbine wheels that incorporate modification AB 173 (Turbomeca S.A. Service Bulletin (SB) No. 283 72 0091) or modification AB 208 (Turbomeca S.A. SB No. 283 72 0117). This AD does not apply to third stage turbine wheels that incorporate Turbomeca S.A. SB No. 283 72 805.

- (1) Third stage turbine wheels, P/N 0 265 25 700 0, all S/Ns;
- (2) Third stage turbine wheels, P/N 0 265 25 702 0, all S/Ns;
- (3) Third stage turbine wheels, P/N 0 265 25 706 0, all S/Ns;
- (4) Third stage turbine wheels, P/N 0 265 25 705 0, with an S/N listed in Appendix 2.1 of Turbomeca S.A. Mandatory Service Bulletin (MSB) No. 283 72 0804, Version D, dated July 24, 2015.

(d) Unsafe Condition

This AD was prompted by a report of a third stage turbine wheel crack detected during engine overhaul. We are issuing this AD to prevent uncontained failure of the third stage turbine wheel, which could result in damage to the engine and damage to the helicopter.

(e) Compliance

Comply with this AD within the compliance times specified, unless already done.

(1) Perform a dye penetrant inspection of the third stage turbine wheel. Use paragraph 2.4.2.2 of Turbomeca S.A. MSB No. 283 72 0804, Version D, dated July 24, 2015, to do the inspection, as follows:

(i) Inspect third stage turbine wheels with 300 engine cycles (EC) or more accumulated since last inspection, or since new, or since last overhaul, or since repair, within 100 EC after the effective date of this AD.

(ii) Inspect third stage turbine wheels with less than 300 EC accumulated since last inspection, or since new, or since last overhaul, or since repair, within 400 EC since last inspection, or since new, or since last overhaul, or since repair.

(2) Repeat the inspection required by this AD within 400 EC since last inspection.

(3) Remove from service any third stage turbine wheels that fail the inspection required by this AD.

(f) Optional Terminating Action

Application of Turbomeca S.A. SB No. 283 72 0805, Version B, dated December 15, 2010 is terminating action for the inspections required by paragraphs (e)(1) and (2) of this AD.

(g) Alternative Methods of Compliance (AMOCs)

The Manager, Engine Certification Office, may approve AMOCs for this AD. Use the procedures found in 14 CFR 39.19 to make your request. You may email your request to: ANE-AD-AMOC@faa.gov.

(h) Related Information

(1) For more information about this AD, contact Brian Kierstead, Aerospace Engineer, Engine Certification Office, FAA, Engine & Propeller Directorate, 1200 District Avenue, Burlington, MA 01803; phone: 781-238-7772; fax: 781-238-7199; email: brian.kierstead@faa.gov.

(2) Refer to MCAI EASA AD 2015-0211, dated October 15, 2015, for related information. You may examine the MCAI in the AD docket on the Internet at <http://www.regulations.gov> by searching for and locating it in Docket No. FAA-2010-0219.

(i) Material Incorporated by Reference

(1) The Director of the Federal Register approved the incorporation by reference (IBR) of the service information listed in this paragraph under 5 U.S.C. 552(a) and 1 CFR part 51.

(2) You must use this service information as applicable to do the actions required by this AD, unless the AD specifies otherwise.

(i) Turbomeca S.A. Mandatory Service Bulletin No. 283 72 0804, Version D, dated July 24, 2015.

(ii) Turbomeca S.A. Service Bulletin No. 283 72 0805, Version B, dated December 15, 2010.

(3) For Turbomeca S.A. service information identified in this AD, contact Turbomeca S.A., 40220 Tarnos, France; phone: (33) 05 59 74 40 00; fax: (33) 05 59 74 45 15.

(4) You may view this service information at FAA, Engine & Propeller Directorate, 1200 District Avenue, Burlington, MA. For information on the availability of this material at the FAA, call 781-238-7125.

(5) You may view this service information at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call 202-741-6030, or go to: <http://www.archives.gov/federal-register/cfr/ibr-locations.html>.

Issued in Burlington, Massachusetts, on June 7, 2016.

Colleen M. D'Alessandro,
Manager, Engine & Propeller Directorate,
Aircraft Certification Service.



2016-12-08 GROB Aircraft AG: Amendment 39-18557; Docket No. FAA-2016-7057; Directorate Identifier 2016-CE-017-AD.

(a) Effective Date

This airworthiness directive (AD) becomes effective July 20, 2016.

(b) Affected ADs

None.

(c) Applicability

This AD applies to Grob Aircraft AG Models G115EG airplanes, serial numbers up to and including 82323/E, certificated in any category.

(d) Subject

Air Transport Association of America (ATA) Code 53: Fuselage.

(e) Reason

This AD was prompted by mandatory continuing airworthiness information (MCAI) originated by an aviation authority of another country to identify and correct an unsafe condition on an aviation product. The MCAI describes the unsafe condition as cracks in the bonded joint of the rear horizontal stabilizer frame. We are issuing this AD to detect and correct cracks in the bonded joint of the rear horizontal stabilizer frame, which if not corrected could propagate into the primary structural elements of the airplane and affect its structural integrity.

(f) Actions and Compliance

Unless already done, do the actions in paragraphs (f)(1) through (6) of this AD.

(1) Within the next 50 hours time-in-service (TIS) after July 20, 2016 (the effective date of this AD), and repetitively thereafter at intervals not to exceed 50 hours, inspect the rear horizontal stabilizer frame following the Accomplishment Instructions in section 1.8, Part A, of GROB Aircraft AG Service Bulletin (SB) No. MSB1078-200, dated February 25, 2016.

(2) If any crack within the green area as defined in Figure 2 of the Accomplishment Instructions in section 1.8, Part A, of GROB Aircraft AG Service Bulletin (SB) No. MSB1078-200, dated February 25, 2016, is found during any inspection required in paragraph (f)(1) of this AD, before further flight, install a temporary placard stating "NO AEROBATICS, NO SPINS AND NO SIDE SLIPS ALLOWED" in full view of the pilot(s) and place a copy of this AD in the airplane flight manual (AFM); and after each day of flight operations, do a crack propagation inspection following the Accomplishment Instructions in Section 1.8, Part B, of GROB Aircraft AG SB No. MSB1078-200, dated February 25, 2016.

(3) If any crack within the red area as defined in Figure 2 of the Accomplishment Instructions in section 1.8, Part A, of GROB Aircraft AG Service Bulletin (SB) No. MSB1078-200, dated February 25, 2016, is found during any inspection required by this AD, before further flight, repair the affected area following the Accomplishment Instructions in Section 1.8, Part C, of GROB Aircraft AG SB No. MSB1078-200, dated February 25, 2016.

(4) Within the next 19 months after July 20, 2016 (the effective date of this AD), unless already done as required by paragraph (f)(3) of this AD, modify the airplane following the Accomplishment Instructions in Section 1.8, Part C, of GROB Aircraft AG SB No. MSB1078-200, dated February 25, 2016.

(5) After modification of the airplane as required by paragraph (f)(3) or (4) of this AD, remove the placard installed as required in paragraph (f)(2) of this AD and remove the copy of this AD from the applicable AFM.

(6) Modification of an airplane as required in paragraph (f)(3) or (4) of this AD, as applicable, constitutes terminating action for the repetitive inspections required in paragraph (f)(1) and (2) of this AD.

(g) Other FAA AD Provisions

The following provisions also apply to this AD:

(1) Alternative Methods of Compliance (AMOCs): The Manager, Standards Office, FAA, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19. Send information to ATTN: Karl Schletzbaum, Aerospace Engineer, FAA, Small Airplane Directorate, 901 Locust, Room 301, Kansas City, Missouri 64106; telephone: (816) 329-4123; fax: (816) 329-4090; email: karl.schletzbaum@faa.gov. Before using any approved AMOC on any airplane to which the AMOC applies, notify your appropriate principal inspector (PI) in the FAA Flight Standards District Office (FSDO), or lacking a PI, your local FSDO.

(2) Airworthy Product: For any requirement in this AD to obtain corrective actions from a manufacturer or other source, use these actions if they are FAA-approved. Corrective actions are considered FAA-approved if they are approved by the State of Design Authority (or their delegated agent). You are required to assure the product is airworthy before it is returned to service.

(h) Related Information

Refer to MCAI European Aviation Safety Agency (EASA) AD No.: 2016-0091, dated May 16, 2016, for related. You may examine the MCAI on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2016-7057.

(i) Material Incorporated by Reference

(1) The Director of the Federal Register approved the incorporation by reference (IBR) of the service information listed in this paragraph under 5 U.S.C. 552(a) and 1 CFR part 51.

(2) You must use this service information as applicable to do the actions required by this AD, unless the AD specifies otherwise.

(i) GROB Aircraft AG Service Bulletin No. MSB1078-200, dated February 25, 2016.

(ii) Reserved.

(3) For GROB Aircraft AG service information identified in this AD, contact GROB Aircraft AG, Product Support, Lettenbachstrasse 9, D-86874 Tussenhausen-Mattsies, Germany, telephone: + 49 (0) 8268-998-105; fax: + 49 (0) 8268-998-200; email: productsupport@grob-aircraft.com; Internet: grob-aircraft.com.

(4) You may view this service information at the FAA, Small Airplane Directorate, 901 Locust, Kansas City, Missouri 64106. For information on the availability of this material at the FAA, call

(816) 329-4148. It is also available on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2016-7057.

(5) You may view this service information that is incorporated by reference at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call 202-741-6030, or go to: <http://www.archives.gov/federal-register/cfr/ibr-locations.html>.

Issued in Kansas City, Missouri, on June 6, 2016.

Robert Busto,
Acting Manager, Small Airplane Directorate,
Aircraft Certification Service.



2016-12-13 Airbus Helicopters (Previously Eurocopter France): Amendment 39-18562; Docket No. FAA-2014-0105; Directorate Identifier 2008-SW-58-AD.

(a) Applicability

This AD applies to Model EC120B helicopters with an engine-to-main gearbox coupling tube assembly (coupling tube), part number (P/N) C631A1101101 or P/N C631A1002101, installed, certificated in any category.

(b) Unsafe Condition

This AD defines the unsafe condition as a crack in a coupling tube. This condition could result in coupling tube failure, loss of engine drive, and a subsequent forced landing of the helicopter.

(c) Affected ADs

This AD supersedes AD 2000-05-17, Amendment 39-11627 (65 FR 13875, March 15, 2000) and AD 2001-04-12, Amendment 39-12131 (66 FR 13232, March 5, 2001).

(d) Effective Date

This AD becomes effective July 27, 2016.

(e) Compliance

You are responsible for performing each action required by this AD within the specified compliance time unless it has already been accomplished prior to that time.

(f) Required Actions

(1) For helicopters with a serial number up to and including 1169, not modified with an improvement of the engine mount in accordance with Eurocopter Service Bulletin (SB) No. 71-003, Revision 1, dated July 18, 2002 (SB 71-003), or not modified by installing a spring-type engine suspension system in accordance with Eurocopter SB No. 71-005, Revision 0, dated May 14, 2004 (SB 71-005), before further flight:

(i) Remove from service the following engine mount parts:

- (A) Support arm, P/N C714A1107201;
- (B) Swaged support arm, P/N C714A1106201;
- (C) Left-hand support bracket, P/N C714A1101102; and
- (D) Right-hand support bracket, P/N C714A1101103.

(ii) Measure the height of the engine mounting base as depicted in Figure 1 of Eurocopter Alert SB No. 04A005, Revision 0, dated July 16, 2003. If the height is more than 10.5 millimeters, replace the engine mount with an engine mount that does not have the parts identified in paragraph (f)(1)(i) of this AD.

(2) For helicopters with a serial number 1170 and larger or helicopters modified with an improvement of the engine mount in accordance with SB 71-003:

(i) Within 25 hours TIS, replace the spring-type engine suspension system and perform a dye-penetrant inspection of the flared coupling for a crack by following the Accomplishment Instructions, paragraphs 2.B.2.a through 2.B.2.c of SB 71-005.

(ii) If there is a crack in the flared coupling, before further flight, replace the coupling with an airworthy coupling.

(3) For helicopters with coupling tube, P/N C631A1002101, installed, before further flight, remove coupling tube, P/N C631A1002101, from service. Do not install coupling tube, P/N C631A1002101, on any helicopter.

(g) Special Flight Permits

Special flight permits may be issued provided there are no cracks in the coupling tube attachment fitting.

(h) Alternative Methods of Compliance (AMOCs)

(1) The Manager, Safety Management Group, FAA, may approve AMOCs for this AD. Send your proposal to: James Blyn, Aviation Safety Engineer, Regulations and Policy Group, Rotorcraft Directorate, FAA, 10101 Hillwood Pkwy., Fort Worth, TX 76177; telephone (817) 222-5110; email 9-ASW-FTW-AMOC-Requests@faa.gov.

(2) For operations conducted under a 14 CFR part 119 operating certificate or under 14 CFR part 91, subpart K, we suggest that you notify your principal inspector, or lacking a principal inspector, the manager of the local flight standards district office or certificate holding district office, before operating any aircraft complying with this AD through an AMOC.

(i) Additional Information

(1) Eurocopter Alert Service Bulletin (ASB) No. 05A003, Revision 2, dated July 16, 2003; Eurocopter ASB No. 05A003, Revision 3, dated May 11, 2004; and Eurocopter Service Bulletin No. 71-003, Revision 1, dated July 18, 2002; which are not incorporated by reference, contain additional information about the subject of this final rule. For Eurocopter service information identified in this final rule, contact Airbus Helicopters, 2701 N. Forum Drive, Grand Prairie, TX 75052; telephone (972) 641-0000 or (800) 232-0323; fax (972) 641-3775; or at <http://www.airbushelicopters.com/techpub>. You may review a copy of the service information at the FAA, Office of the Regional Counsel, Southwest Region, 10101 Hillwood Pkwy., Room 6N-321, Fort Worth, TX 76177.

(2) The subject of this AD is addressed in Direction Generale de L'Aviation Civile (DGAC) AD No. F-2003-325 R1, Revision 1, dated May 12, 2004. You may view the DGAC AD on the Internet at <http://www.regulations.gov> in Docket No. FAA-2014-0105.

(j) Subject

Joint Aircraft Service Component (JASC) Code: 6310, Engine/Transmission Coupling–Coupling Tube, Engine Mount, and Engine Mount Base.

(k) Material Incorporated by Reference

(1) The Director of the Federal Register approved the incorporation by reference of the service information listed in this paragraph under 5 U.S.C. 552(a) and 1 CFR part 51.

(2) You must use this service information as applicable to do the actions required by this AD, unless the AD specifies otherwise.

(i) Eurocopter Alert Service Bulletin No. 04A005, Revision 0, dated July 16, 2003.

(ii) Eurocopter Service Bulletin No. 71-005, Revision 0, dated May 14, 2004.

(3) For Eurocopter service information identified in this final rule, contact Airbus Helicopters, 2701 N. Forum Drive, Grand Prairie, TX 75052; telephone (972) 641-0000 or (800) 232-0323; fax (972) 641-3775; or at <http://www.airbushelicopters.com/techpub>.

(4) You may view this service information at FAA, Office of the Regional Counsel, Southwest Region, 10101 Hillwood Pkwy., Room 6N-321, Fort Worth, TX 76177. For information on the availability of this material at the FAA, call (817) 222-5110.

(5) You may view this service information that is incorporated by reference at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call (202) 741-6030, or go to: <http://www.archives.gov/federal-register/cfr/ibr-locations.html>.

Issued in Fort Worth, Texas, on June 9, 2016.

Scott A. Horn,
Acting Manager, Rotorcraft Directorate,
Aircraft Certification Service.



2016-13-04 BRP-Powertrain GmbH & Co KG (formerly BRP-Rotax GmbH & Co KG, Bombardier-Rotax GmbH & Co. KG, and Bombardier-Rotax GmbH): Amendment 39-18568; Docket No. FAA-2016-2042; Directorate Identifier 2016-NE-02-AD.

(a) Effective Date

This AD becomes effective July 27, 2016.

(b) Affected ADs

None.

(c) Applicability

This AD applies to BRP-Powertrain GmbH & Co KG Rotax model 912 F2, 912 F3, 912 F4, 912 S2, 912 S3, 912 S4, 914 F2, 914 F3, and 914 F4 reciprocating engines with a cylinder head that has a part number (P/N) listed in Figure 1 to paragraph (c) of this AD and that is installed in position 2 or 3.

Figure 1 to Paragraph (c) of This AD—Post-Modification Cylinder Head P/N

Engine model	Cylinder head P/N
912 F2, 912 F3, 912 F4, 914 F2, 914 F3, and 914 F4	P/N 413235 or P/N 413236.
912 S2, 912 S3, and 912 S4	P/N 413185.

(d) Reason

This AD was prompted by a design change introduced by the manufacturer that relocated the engine cylinder head temperature sensor to a new location and converted it to a coolant temperature sensor. We are issuing this AD to prevent exceeding coolant temperature limits, which could result in loss of engine coolant, damage to the engine, and loss of control of the airplane.

(e) Actions and Compliance

Comply with this AD within 6 months after the effective date of this AD, unless already done.

(1) For engines with cylinder heads that have a P/N listed in Figure 1 to paragraph (c) of this AD installed on both position 2 and position 3, change the engine model designation on the engine type data plate to include a "-01" suffix. Use paragraph 3.1.1 of BRP-Powertrain Service Bulletin (SB) SB-912-068/SB-914-049, dated April 16, 2015, to make this change.

(2) For engines with only one cylinder head having a P/N listed in Figure 1 to paragraph (c) of this AD installed in position 2 or 3, do one of the following:

(i) Replace the cylinder head having a P/N listed in Figure 1 to paragraph (c) of this AD with a P/N 623682 cylinder head on Rotax 912 F2, 912 F3, 912 F4, 914 F2, 914 F3, and 914 F4 engines and

with a P/N 623687 cylinder head on Rotax 912 S2, 912 S3, and 912 S4 engines. If you complete the actions in paragraph (e)(2)(i), no further action is required. Or,

(ii) Install cylinder heads identified in Figure 1 to paragraph (c) of this AD on both cylinder head positions 2 and 3 and change the engine model designation of the engine type data plate in accordance with paragraph (e)(1) of this AD.

(3) For engines re-identified in accordance with paragraph (e)(1) or (e)(2)(ii) of this AD, before further flight, modify the aircraft cockpit instrumentation and related documentation to indicate a maximum coolant temperature limit of 120 degrees Celsius using FAA-approved procedures. These re-identified engines remain eligible for installation on approved aircraft-engine combinations.

(f) Alternative Methods of Compliance (AMOCs)

The Manager, Engine Certification Office, FAA, may approve AMOCs for this AD. Use the procedures found in 14 CFR 39.19 to make your request. You may email your request to: ANE-AD-AMOC@faa.gov.

(g) Related Information

(1) For more information about this AD, contact Robert Green, Aerospace Engineer, Engine Certification Office, FAA, Engine & Propeller Directorate, 1200 District Avenue, Burlington, MA 01803; phone: 781-238-7754; fax: 781-238-7199; email: robert.green@faa.gov.

(2) For more information about the installation modifications described in paragraph (e)(3) of this AD, contact Jim Rutherford, Aerospace Engineer, FAA, Small Airplane Directorate, 901 Locust Ave. Room 301, Kansas City, MO; phone: 816-329-4165; fax: 816-329-4090; email: Jim.Rutherford@faa.gov.

(3) Refer to MCAI European Aviation Safety Agency, AD 2015-0240, dated December 18, 2015, for more information. You may examine the MCAI in the AD docket on the Internet at <http://www.regulations.gov> by searching for and locating it in Docket No. FAA-2016-2042.

(4) The following aircraft service information, which are not incorporated by reference in this AD, contain FAA-approved procedures for complying with paragraph (e)(3) of this AD and can be obtained from BRP-Powertrain GmbH & Co. KG, using the contact information in paragraph (h)(3) of this AD:

Figure 2 to Paragraph (g) of This AD—Aircraft Type/Model and Service Information

Type/model(s)	SB
Aquila AT01	SB-AT01-029.
TECNAM P92, P2002 and P2006T	SB-183-CS.
TECNAM P2008 JC	SB-185-CS.
Diamond H 36 “Dimona” and HK 36 “Super Dimona”	OSB 36-111.
Diamond DV 20 “Katana”	OSB 20-066.
Diamond (Canada) DA20-A1 “Katana”	SB Da20-72-04.
M&D AVO 68 “Samburo”	TM 808-31.
Scheibe SF 25 C and SF 36 R	SI_02-14.

(h) Material Incorporated by Reference

(1) The Director of the Federal Register approved the incorporation by reference (IBR) of the service information listed in this paragraph under 5 U.S.C. 552(a) and 1 CFR part 51.

(2) You must use this service information as applicable to do the actions required by this AD, unless the AD specifies otherwise.

(i) BRP-Powertrain Service Bulletin SB-912-068/SB-914-049 (one document), dated April 16, 2015.

(ii) Reserved.

(3) For BRP-Powertrain service information identified in this AD, contact BRP-Powertrain GmbH & Co. KG, Rotaxstrasse 1, A-4623 Gunskirchen, Austria; phone: +43 7246 6010; fax: +43 7246 601 9130; email: airworthiness@brp.com; Internet: www.flyrotax.com.

(4) You may view this service information at FAA, Engine & Propeller Directorate, 1200 District Avenue, Burlington, MA. For information on the availability of this material at the FAA, call 781-238-7125.

(5) You may view this service information at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call 202-741-6030, or go to: <http://www.archives.gov/federal-register/cfr/ibr-locations.html>.

Issued in Burlington, Massachusetts, on June 14, 2016.

Colleen M. D'Alessandro,
Manager, Engine & Propeller Directorate,
Aircraft Certification Service.