This data sheet which is part of type certificate No. A12CE prescribes conditions and limitations under which the product for which the type certificate was issued meets the airworthiness requirements of the Federal Aviation Regulations.

Type Certificate Holder: Beechcraft Corporation
10511 East Central
Wichita, KS 67206

Type Certificate Holder Record:
Beech Aircraft Company transferred to Raytheon Aircraft Company on April 15, 1996
Raytheon Aircraft Company transferred to Hawker Beechcraft Corporation on March 26, 2007
Hawker Beechcraft Corporation transferred to Beechcraft Corporation on April 12, 2013

I. Model 60, Duke, (Normal Category), Approved February 1, 1968
Model A60, Duke, (Normal Category), Approved January 30, 1970
Model B60, Duke, (Normal Category), Approved October 5, 1973

Engines: Lycoming TIO-541-E1A4 or TIO-541-E1C4 (2 of either or 1 of each for S/N P-4 through P-522)
2 Lycoming TIO-541-E1C4 (for S/N P-523 and up)

Fuel: 100LL or 100 minimum grade aviation gasoline
115/145 alternate grade aviation gasoline

Oil: Ashless dispersant multi-grade conforming to MIL-L-22851 or a Lycoming approved synthetic oil

Engine limits: Takeoff and maximum continuous power, 2900 r.p.m. at 41.5 in. hg., 380 b. hp.
Maximum normal operating power, 2750 r.p.m. at 36.5 in. hg., 301 hp.
(for S/N P-523 and up)

Propeller and propeller limits: (a) 2 (in any combination) Hartzell three-blade propellers
(No further reduction permitted)
Pitch settings at 30 in. sta.:
low 13°-14°, high 81.7°
HC-F3YR-2/C7479-2R
or HC-F3YR-2/C7479B-2R
or HC-F3YR-2F/FC7479-2R
or HC-F3YR-2F/FC7479B-2R
or HC-F3YR-2UF/FC7479-2R
or HC-F3YR-2UF/FC7479B-2R
I. **Model 60, Model A60, Model B60** (cont’d)

or HC-F3YR-2UF/FC7479K-2R (for S/N P-579 and up)

See NOTE 6

(b) Beech 60-389000 governor

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### Airspeed limits

- **(IAS)**
  - Never exceed: 268 m.p.h. (233 knots)
  - Maximum structural cruising: 238 m.p.h. (207 knots)
  - Maneuvering: 184 m.p.h. (160 knots)
  - Maximum flap extension speed:
    - Approach position 15°: 200 m.p.h. (174 knots)
    - Full down position 30°: 154 m.p.h. (134 knots) (60 and A60)
    - 161 m.p.h. (140 knots) (B60)
  - Landing gear extended: 200 m.p.h. (174 knots)
  - Landing gear operating: 200 m.p.h. (174 knots)

### C.G. range

- **(Landing Gear Extended)**
  - (+134.2) to (+139.2) at 6725 lb. Model 60 (See NOTES 4 and 5)
  - (+134.6) to (+139.2) at 6775 lb. Models A60 and B60 (See NOTE 5)
  - (+128.0) to (+139.2) at 5100 lb. or less (All Models)

Straight line variation between points given

Moment change due to retracting landing gear (+857 in.-lb.)

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![Graph](https://via.placeholder.com/150)

### Empty weight C.G. range

None

### Maximum weight

- Takeoff and landing: 6725 lb. (Model 60) (See NOTES 4 & 5)
- 6775 lb. (Models A60/B60) (See NOTE 5)
- Ramp weight: 6819 lb.

### Minimum Crew

One pilot

### No. of seats and Cargo loading

Maximum 6 including crew at +141. See loading instructions in Pilot’s Operating Manual for approved seating and cargo configurations.

### Maximum baggage

- 500 lb. at +75 (nose compartment)
- 315 lb. at +230 (aft cabin area)
### I. Model 60, Model A60, Model B60 (cont’d)

<table>
<thead>
<tr>
<th>Fuel capacity</th>
<th>Tank</th>
<th>Cap. Gal.</th>
<th>Usable Gal. (Full Fuel Only)</th>
<th>Code #</th>
<th>Arm</th>
</tr>
</thead>
<tbody>
<tr>
<td>(P-4 through P-195)</td>
<td>L &amp; R Wing or L &amp; R Wing &amp; Nacelle</td>
<td>73.5 ea. or 103.5 ea.</td>
<td>71 ea. or 102 ea.</td>
<td>(1)</td>
<td>137.9 or 139.0</td>
</tr>
<tr>
<td></td>
<td>L &amp; R Wing &amp; Nacelle</td>
<td>103.5 ea.</td>
<td>96 ea.</td>
<td>(2)</td>
<td>139.5</td>
</tr>
<tr>
<td>(P-196 through P-219)</td>
<td>L &amp; R Wing or L &amp; R Wing &amp; Nacelle</td>
<td>103.5 ea. or 103.5 ea.</td>
<td>101 ea. or 102 ea.</td>
<td>(4)</td>
<td>139.0 or 139.0</td>
</tr>
<tr>
<td>(P-220 through P-364 except P-348)</td>
<td>L &amp; R Wing or L &amp; R Wing &amp; Nacelle</td>
<td>103.5 ea.</td>
<td>101 ea.</td>
<td>(5)</td>
<td>139.0</td>
</tr>
<tr>
<td>(P-348, P-365 and after)</td>
<td>L &amp; R Wing or L &amp; R Tip, Wing &amp; Nacelle</td>
<td>103.5 ea. or 118.5 ea.</td>
<td>101 ea. or 116 ea.</td>
<td>(5)</td>
<td>139.0 or 139.7</td>
</tr>
</tbody>
</table>

**Code # Explanation:**

*#1* As manufactured with unbaffled tanks.

*#2* Unbaffled tanks after compliance with S.I. 0559-281.

*#3* Baffled tanks after compliance with S.I. 0559-281.

*#4* As manufactured with baffled tanks.

*#5* As manufactured with baffled tanks and increased unusable fuel requirement.

See NOTE 1 for data on unusable fuel

- **Oil capacity (Wet Sump):** 26 qt. (+88)
- **Max. Operating Limit:** 30,000 ft. pressure altitude

**Control surface movements**

<table>
<thead>
<tr>
<th>Movement</th>
<th>Maximum</th>
<th>(LH)</th>
</tr>
</thead>
<tbody>
<tr>
<td>Wing flaps</td>
<td>30°</td>
<td></td>
</tr>
<tr>
<td>Aileron</td>
<td>Up 25°</td>
<td>Down 15°</td>
</tr>
<tr>
<td>Aileron tab (LH only)</td>
<td>Up 10°</td>
<td>Down 10°</td>
</tr>
<tr>
<td>Aileron tab anti-servo</td>
<td>Up 12°</td>
<td>Down 7°</td>
</tr>
<tr>
<td>Elevator</td>
<td>Up 17°</td>
<td>Down 15°</td>
</tr>
<tr>
<td>Elevator tab (LH only)</td>
<td>Up 10°</td>
<td>Down 30°</td>
</tr>
<tr>
<td>Elevator tab servo</td>
<td>Up 6°</td>
<td>Down 7°</td>
</tr>
<tr>
<td>Rudder</td>
<td>Right 33°</td>
<td>Left 28°</td>
</tr>
<tr>
<td>Rudder tab</td>
<td>Right 20°</td>
<td>Left 20°</td>
</tr>
</tbody>
</table>

**Serial Nos. eligible**

- Model 60: P-4 through P-126 (except P-123)
- Model A60: P-123, P-127 through P-246
- Model B60: P-247 and up
Data Pertinent to All Models

Datum
Located 100 in. forward of front pressure bulkhead

Leveling means
Drop plumb line between leveling screws in cabin door frame rear edge.

Certification basis
Part 23 of the Federal Aviation Regulations effective February 1, 1965, as amended by 23-1, 23-2, 23-3; and Paragraph 23.959 of Amendment 23-7 for S/N P-220 and up (also S/N P-4 through P-219 after compliance with S.I. 0559-281); and Amendment 23-11 for S/N P-402 and up (also S/N P-4 through P-401 when modified per Beech Aircraft Corporation Kit Drawing 60-3006); and Paragraphs 23.1385(c), 23.1387(a), 23.1387(e) of Amendment 23-12; Part 36, as amended by 36-1 through 36-10 for S/N P-523 and up; and Special Conditions dated May 16, 1967, forwarded with FAA letter dated June 1, 1967; approved for flight into known icing conditions when equipped as specified in the Approved Airplane Flight Manual.

Equivalent safety findings: FAR 23.75; 23.175(b); 23.621 (S/N P-4 through P-592); 23.1305(g); 23.1545(a) and 23.1583(a) (S/N P-486 and up) (also S/N P-247 through P-485 when modified per Beech Aircraft Corporation Kit Drawing 60-5023); 23.1549 and 23.1563(a).

Application for Type Certificate dated December 22, 1956; Type Certificate No. A12CE issued February 1, 1968, obtained by the manufacturer under delegation option procedures.

Production basis
Production Certificate No. 8 issued and Delegation Option Manufacturer No. CE-2 authorized to issue airworthiness certificates under delegation option provisions of Part 21 of the Federal Aviation Regulations.

Equipment
The basic required equipment as prescribed in the applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification.

In addition:

1. For flights into known icing conditions, these flight manual supplements and the equipment noted therein:
   60-590001-17 Flight into known icing conditions; revision dated February 3, 1978, or later revision
   60-590001-11 Continuous pressure operated surface deice system
   60-590001-13 Goodrich electrothermal propeller deice system Equipment (Cont.)

2. For all other operations:
   Pre-stall warning indicator P/N 151-6, 151-7, 190-2 or 190-3 with mounting plate 151-202-1 or 151-202-2 (Safe Flight Corp.) or 191-52 assembly.

3. Model A60 Airplanes (S/N P-144 through P-246) require Airplane Flight Manual P/N 60-590000-5E, revision E-6 dated November 6, 1974, or later revision.
   Model B60 Airplanes (S/N P-247 and up) require Airplane Flight Manual P/N 60-590000-11, revision A-5 dated November 1, 1974, or later revision.
Data Pertinent to All Models (cont’d)

NOTE 1. Current weight and balance data including list of equipment included in certificated empty weight and loading instructions when necessary must be provided for each aircraft at the original certification.

   (a) The certificated empty weight and corresponding center of gravity locations must include unusable fuel as follows:

   (Refer to fuel capacity information for explanation of Code # applicability):

   Code #1 and #4 Systems          24 lb.  (+135)
   Code #2 System                  90 lb.  (+131.7)
   Code #3 or #5 Systems           30 lb.  (+134)

   (b) The basic empty weight and corresponding center of gravity must include oil of 49 lbs. at +88.

NOTE 2. The following placard must be displayed in front of and in clear view of the pilot:

"This airplane must be operated in the normal category in compliance with the operation limitations stated in the form of placards, markings and manuals."


NOTE 4. Model 60 (S/N P-4 through P-126 except P-123) when modified to Beech Dwg. 60-5008 or equipped with Airplane Flight Manual 60-590000-5E dated November 6, 1974, or later revision eligible for a maximum of 6775 lb.

NOTE 5. For aircraft equipped with 60-810012-15 (LH) or 60-810012-16 (RH) shock absorbers and 10 PR tires, the landing weight is 6775 lbs. For 8PR tires and 60-810012-13 (LH) and -14 (RH) or lower dash number shock absorbers, a landing weight of 6600 lbs. must be observed. For 10 PR tires and 60-810012-13 (LH) and -14 (RH) or lower dash number shock absorbers, a landing weight of 6450 lbs. must be observed. For 8PR tires and 60-810012-15 (LH) or -16 (RH) or higher dash number shock absorbers, a landing weight of 6600 lbs. must be observed.

NOTE 6. Propeller hub assemblies with the "U" designation shall be installed in pairs only.

Contact Beech Aircraft Corporation as necessary to obtain availability information concerning the drawings and kits which are referenced by this publication.

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