United States of America
Department of Transportation_Federal Aviation Administration

Supplemental Type Certificate

Number SA01672SE

This certificate, issued to Northwest Turbine LLC
E. 6427 Rutter Road, Suite D
Spokane, WA 99212

certifies that the change in the type design for the following product with the limitations and conditions therefore as specified hereon meets the airworthiness requirements of Part 23 of the Federal Aviation Regulations.

Original Product—Type Certificate Number: A12CE
Make: Raytheon Beech
Model: B-60

Description of the Type Design Change: Installation of two Pratt & Whitney PT6A-21 or two Pratt & Whitney PT6A-35 engines, two Hartzell HC-E4N-3N/D8292B-2 propellers and associated hardware in accordance with Rocket Engineering Master Drawing List 600.00.000, Rev H, dated March 31, 2007, or later FAA-approved revision.

Limitations and Conditions: Modification of the above model airplane requires concurrent or prior incorporation of STC SA00370SE (Strakes), STC SA5761NM (Vortex Generators), and STC SA00112SE (Winglets). Airplanes modified in accordance with this STC must be operated in the Normal Category. Airplanes modified in accordance with this STC must be maintained in accordance with Northwest Turbine, LLC document 600.101.DOC.

(continued on pages 3, 4, and 5)

This certificate and the supporting data which is the basis for approval shall remain in effect until surrendered, suspended, revoked, or a termination date is otherwise established by the Administrator of the Federal Aviation Administration.

Date of application: December 15, 2000
Date of issuance: May 12, 2006

Date reissued: May 11, 2007
Date amended: November 20, 2008

By direction of the Administrator

________________________________________ (Signature)

Acting Manager, Seattle Aircraft Certificate Office

(Title)

Any alteration of this certificate is punishable by a fine of not exceeding $1,000, or imprisonment not exceeding 3 years, or both.

FAA Form 8110-2(10-68)

This certificate may be transferred in accordance with FAR 21.47.
Limitations and Conditions: (cont’d)

Additionally, approval of this change in type design applies to the above model aircraft only. This approval should not be extended to other aircraft of this model on which other previously approved modifications are incorporated unless it is determined that the relationship between this change and any of those other previously approved modifications, including changes in type design, will introduce no adverse effect upon the airworthiness of that aircraft. A copy of this certificate, dated November 20, 2008 and the Aircraft Flight Manual Supplement, dated May 12, 2006 or later FAA-approved revision, must be maintained as part of the permanent records of the modified aircraft.

If the holder agrees to permit another person to use this certificate to alter the product, the holder shall give the other person written evidence of that permission.

AIRPLANE LIMITS
Altitude Limit.................................................................................. 28000 ft

Maximum Weights
Maximum Ramp Weight.................................................................. 7039 Lb
Maximum Takeoff Weight ............................................................. 7000 Lb
Maximum Landing Weight ........................................................... 6775 Lb
Maximum Zero Fuel Weight ......................................................... 5900 Lb

ENGINE LIMITS
Number of Engines........................................................................ 2
Engine Manufacturer..................................................................... Pratt & Whitney
Engine Model............................................................................... PT6A-35, or PT6A-21(The same model engine must be installed in both positions.)
Takeoff Torque (PT6A-21 and PT6A-35)....................................... 1260 ft-lb @ 2190 RPM

Maximum Gas Generator Speed (%) for both PT6A-21 and PT6A-35
Continuous .................................................................................. 101.5
Acceleration (Two second limit) .................................................. 102.6
Reverse ...................................................................................... 101.5

Maximum Interturbine Temp (ITT, deg C)

<table>
<thead>
<tr>
<th></th>
<th>PT6A-35</th>
<th>PT6A-21</th>
</tr>
</thead>
<tbody>
<tr>
<td>Takeoff</td>
<td>805</td>
<td>695</td>
</tr>
<tr>
<td>Maximum Climb/Maximum Cruise</td>
<td>770</td>
<td>680</td>
</tr>
<tr>
<td>Starting (Two second limit)</td>
<td>1090</td>
<td>1090</td>
</tr>
<tr>
<td>Acceleration (Two second limit)</td>
<td>880</td>
<td>825</td>
</tr>
<tr>
<td>Reverse</td>
<td>805</td>
<td>695</td>
</tr>
<tr>
<td>Idle</td>
<td>685</td>
<td>660</td>
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</table>

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Northwest Turbine LLC
Issued: May 12, 2006
Reissued:

Limitations and Conditions: (cont'd)

PROPELLER LIMITS
   Number of Propellers .................................................... 2
   Propeller Manufacturer ................................................... Hartzell
   Blade Model ................................................................. D8292(B)-2
   Number of Blades ........................................................... 4
   Hub Model ................................................................. HC-E4N-3N
   Propeller Diameter (Inches) ............................................. 80.0 Min, 80.5 Max.

FUEL
   Approved Fuels ........................................................... JP4, JP5, JET A, JET A-1, JET B
   Fuel Temperature Limit ............................................... 100 deg F

Fuel Capacity (U.S. Gallons)

<table>
<thead>
<tr>
<th></th>
<th>Total</th>
<th>Usable</th>
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<tbody>
<tr>
<td>Left Wing Tank</td>
<td>138.4</td>
<td>135.9</td>
</tr>
<tr>
<td>Right Wing Tank</td>
<td>132.5</td>
<td>130.0</td>
</tr>
<tr>
<td>Totals</td>
<td>270.9</td>
<td>265.9</td>
</tr>
</tbody>
</table>

Total unusable fuel (5 gallons) must be added to the empty weight of the aircraft.

Note 1: The Certification Basis for this modification is FAR Part 23, effective February 1, 1965, as amended by 23-1, 23-2, 23-3, and paragraph 23.959 of Amendment 27. Paragraphs 23.1385(c), 23.1387(a), 23.1387(e) of Amendment 23-12, Part 36, latest Amendment, and special Conditions, dated May 16, 1967 forwarded with FAA letter, dated June 1, 1967.

Additionally the applicant has volunteered compliance with the following:


Any alteration of this certificate is punishable by a fine of not exceeding $1,000, or imprisonment not exceeding 3 years, or both.

FAA Form 8110-2-1 (10-69) This certificate may be transferred in accordance with FAR 21.47.
Northwest Turbine LLC
Issued: May 12, 2006
Reissued: 

Limitations and Conditions: (cont’d)

Note 2: This airplane/engine/propeller combination is life limited. Incorporation of this STC does not affect that life limit.

Note 3: Aircraft modified in accordance with this Supplemental Type Certificate have been shown by test to comply with the requirements of FAR Part 36 Appendix G.

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